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HYPERSONIC RESEARCH ENGINE/AEROTHERMODYNAMIC

INTEGRATION MODEL - EXPERIMENTAL RESULTS

Volume III - Mach 7 Component Integration and Performance

by

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and

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(Contract No. NAS1-6666)

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SUMMARY

An extensive aerothermodynamic development program for the purpose of advancing the technology of airbreathing propulsion for hypersonic flight has been conducted by NASA in the form of the Hypersonic Research Engine (HRE) Project. The engine components (inlet, combustor, and nozzle) aerothermodynamic development program culminated in the testing of an engine which integrated these components and allowed assessment of engine performance at Mach numbers of 5, 6, and 7. This engine was termed the Aerothermodynamic Integration Model (AIM) and was a water-cooled, hydrogen-fueled, full-scale configuration of the HRE design concept, 18 inches in diameter at the cowl lip and 87 inches long.

Descriptions of the AIM tests and a computer program used in the engine performance analyses, as well as data results and analyses, have been previously documented. All of the results of the engine performance computer program, including enough information to enable additional analysis or interpretation of the data, are reported in four volumes. Volume I presents Mach 6 component integration results that were obtained with supersonic combustion. During the integration tests, inlet unstart limits were determined for fuel injection from the first stage fuel injectors only and for multi-stage fuel injection. Optimization of the fuel injector combination that would yield the best combustion and engine performance was attempted. Volume II presents Mach 6 engine performance results during supersonic and subsonic combustion modes. Combustion mode transition was successfully performed, exit surveys made, and effects of altitude, angle of attack, and inlet spike position were determined during these tests. Volume III (present report) presents Mach 7 component integration and engine performance results with supersonic combustion modes. Fuel injector optimization was again attempted, exit surveys made, and the effects of low free-stream total temperature, free-stream oxygen content, and angle of attack were studied during these tests. Volume IV presents Mach 5 component integration and engine performance results with supersonic and subsonic combustion modes. Combustion mode transition was successfully demonstrated, exit surveys made, and effects of free-stream total temperature, free-stream oxygen content, and angle of attack were investigated during these tests.

INTRODUCTION

The NASA Hypersonic Research Engine (HRE) Project was undertaken to design, develop, and construct a hypersonic research ramjet engine for high performance and to flight test the developed concept on the X-15-2A airplane over the speed range from Mach 3 to 8. It should be emphasized that from the beginning the design was specified to be a research ramjet engine to conduct meaningful experiments and was in no sense intended to be a small-scale prototype of a propulsion system for any particular mission.

About one year after the development phase of the HRE program was underway, the X-15 program was phased out; as a result, adjustments to the project plan and scope were necessitated, which were, however, effected without detriment to achievement of the basic project objectives. The result of the adjustment was that ground testing became the major experimental effort for the HRE program. Engine aerothermodynamic components (inlet, combustor, and nozzle) were developed in separate ground-test programs. Results of the development tests are documented in references 1 through 3. Regeneratively cooled engine structures were also included in the ground-testing program. Tests of the hydrogen-cooled engine structure progressed from small panels and problem area components in laboratory setups to wind-tunnel tests at Mach 6.7 of a full-scale, flight-weight engine termed the Structure Assembly Model (SAM). Results of this program, which was completed in May 1971, are reported in reference 4. Culmination of all the HRE development testing was the engine tests of what was termed the Aerothermodynamic Integration Model (AIM). The purpose of the tests of this full-scale, watercooled, hydrogen-fueled engine was to integrate the aerothermodynamic components and to assess the engine performance at Mach numbers of 5, 6, and 7. Successful tests of the AIM were completed in April 1974.

The AIM employed the HRE design concept of an axisymmetric engine, 18 inches in diameter at the cowl lip and 87 inches long. Versatility was incorporated into the AIM to allow: (1) inlet spike translation for optimum air flow and inlet internal contraction ratio variation; and (2) hydrogen fuel injection for tailored fuel distribution for proper heat release in a diverging combustor, and to change the mode of combustion from supersonic to subsonic or vice versa. The AIM tests are reported in reference 5 and data results of the tests have been analyzed in terms of engine performance by use of a computer program (ref. 6) generated during the HRE program. Results of these analyses are reported in references 7 through 9.

The purpose of the present reports (herein and refs. 10 to 12) is to present experimental engine performance results obtained from computer program analyses of the test data. These results contain the free-stream conditions, pressure distributions, fuel injection configuration and rate, etc., that should enable additional analysis or interpretation of results other than those previously reported. It



should be noted that all units are in U.S. Customary Units because the data results from the HRE contracts, which were initiated in May 1965 with a follow-on effort in February 1967, were under that system. Because of the cost that would have been incurred if the contractors had been required to change to the metric system, the U.S. Customary Units were retained through the HRE contractual effort; this procedure is consistent with the guidelines for conversion established by NASA.

SYMBOLS

All units are in U.S. Customary Units because of the reason noted above.

A area, ft.²

M Mach number

P or p pressure, psia

r radius, in.

 R_{Cl} cowl lip radius at 120 tangent point (see table 3), in.

x longitudinal distance from inlet spike virtual tip (see table 3), in.

longitudinal distance from inlet spike virtual tip to the cowl lip 120 tangent point (see table 3), in.

Δx longitudinal distance inlet centerbody moved from inlet physical close-off, in.

 $\Delta\Delta x$ difference between an actual x_{CL} value and the Mach 6 x_{CL} value of 34.884 in., in.

T temperature, OR

 α angle of attack, deq.

fuel equivalence ratio; value of unity is for stoichiometric combustion (subscript symbols or notations, such as ϕ_{1A} or ER1A, represent the values for the designated fuel injector (e.g., 1A), EROA is the sum of all ϕ -values).

Subscripts:

0 free stream

ref. reference condition

th throat

T total

APPARATUS

Experimental Tests

Experimental tests of the HRE/AIM were conducted in the Plum Brook Hypersonic Tunnel Facility (HTF) (figs. 1(a) and 1(b)) at nominal Mach numbers of 5, 6, and 7. The AIM is shown partially installed in the HTF in the photographs of figures 1(c) and 1(d). During the tests the engine was nearly completely enshrouded except for an 11-inch gap between the facility nozzle exit and the front of the shroud as depicted in the schematic of figure 1(e). This test configuration was suggested by results of a subscale tunnel starting investigation reported in reference 13.

A description of the facility and the results of calibration tests are presented in reference 14. The test facility used an induction-heated, drilled-core graphite storage bed to raise the temperature of nitrogen to a nominal 4960° R at a maximum design pressure of 1200 psia. The nitrogen was mixed with ambient-temperature oxygen to produce synthetic air. Diluent nitrogen was added with the oxygen in the mixture at tunnel Mach numbers below 7 to control free-stream total temperature and to supply the correct weight flow. Because of facility heater deterioration and a lack of time to implement necessary repairs, true temperature simulation of 3700° R at Mach 7 was not achieved; a maximum temperature of about 3100° R was obtained.

The original test plan is summarized in table 1. Because of testing problems and limitations in facility schedule, the test plan was altered to provide a maximum of data to meet the test objectives. Details of the AIM tests are described in reference 5. General test conditions, results, and remarks of the AIM tests were tabulated in references 5 and 9 and are presented herein as table 2. All tests (reading numbers in second column) are listed including the tests that were aborted because of tunnel starting or other problems. Run numbers were assigned to AIM reading numbers or groups of AIM reading numbers with the same test objective (some readings represent zero success, partial success, or are reruns of others) to provide a means for a cross-check with the original plan.

Model

The HRE/AIM was a full-scale (18 inches in diameter at the cowl and 87 inches long), water-cooled, hydrogen-fueled research engine. Details of the design and fabrication of the AIM have been reported in references 16 through 29. The design is described generally in references 5 and 9, and some difficulties encountered with the AIM during the tests are discussed in reference 5

A schematic of the AIM is presented in figure 2 and the coordinates are listed in table 3. The AIM incorporated a mixed compression inlet with a translating spike that enabled the close-off of the engine (an early HRE program

requirement). The inlet was designed for spike translation to the most open position for Mach 4 to 6 operation with spillage occurring up to Mach 6. At Mach 6 "shock-on-lip" occurred, and from Mach 6 to 8 the spike was designed to translate to maintain shock-on-lip over this Mach number range. An "upsloping throat" was incorporated in the inlet which enabled the inlet to not only maintain shock-on-lip with spike translation for Mach 6 to 8, but also to have increased inlet contraction ratio with increased Mach number. The combustor was designed with diverging walls and the area distribution is shown in figure 3(a) with fuel injector locations indicated. Figure 3(b) presents a sketch of the combustor with the locations of the staged fuel injectors and two sets of ignitors indicated (a third set of ignitors planned for the outerbody at an x-station of 54.38 inches was not installed). The set of ignitors at an x-station of 42.0 inches malfunctioned and use was discontinued (see fig. 3(b)) about midway in the Mach 6 test program (see discussion in ref. 5). Injectors 1A, 1B, 1C, 4, 2A, and 2C were designed to allow optimum distribution of the fuel in the combustor to obtain a fuel equivalence ratio, ϕ , of unity during the supersonic combustion mode. During the supersonic combustion mode, it was desired to inject the maximum amount of fuel from the first-stage injectors (IA and IB) without unstarting the inlet; all of the fuel was designed to be injected from injectors 1A and 1B at Mach 8. Injectors 3A and 3B were designed for use in the subsonic combustion mode. The locations are tabulated in figure 3(b) for the designed Mach 6 inlet operating position; cowl lip positions other than the Mach 6 position (because of spike translation) result in different x-station values for the injectors and ignitors on the outer wall and also for injector 3B. These changes are accounted for in the performance results presented herein.

Instrumentation

Planned instrumentation for the AIM is documented in reference 15. All of the instrumentation planned was not used because of facility instrumentation recording channel limitations or damages to instrumentation in inaccessible places during the AIM final assembly or during AIM repairs at the test site. A list of all planned instrumentation is presented in table 4 (obtained from ref. 5) with notations indicating the items not installed or damaged, the recording channel numbers for each item used, and the ranges of the pressure transducers or thermocouples.

Method of Calculation

A computer program that incorporated methods described in reference 15 was used in reducing the data from the AIM tests to engineering units. Listings of this program were checked for accuracy and determination of steady-state conditions. Times of interest were selected from each run and the information from the engineering units computer program was used in a performance analysis computer program which incorporated methods described in reference 6. After the erroneous surface pressures were eliminated, the remaining pressures at each station were averaged by the performance computer program which then performed surface-pressure integration by linear interpolation and determined the skinfriction coefficients. Chemical equilibria of the synthetic air and fuel-air mixtures were calculated by the program using methods described in reference 30.

Description of Performance Program Methods

General.- Several methods were used to establish validity of critical parameters, such as the wind tunnel Mach number. The first method used curves generated from instrumentation rakes installed during calibration of the wind tunnel. The second method used measured values of wind tunnel total pressure and temperature, and pitot pressure at the spike tip along with real-gas, normal-shock solution to calculate the wind tunnel Mach number. The third method used measured values of wind tunnel total temperature, spike-tip pitot pressure, and spike cone surface pressure, along with the real-gas, normal-and conical-shock solutions, to calculate the wind tunnel Mach number. Calculations made utilizing each of the three methods indicated good agreement. After confidence was established in the three methods, the use of the third method was discontinued, since it required excessive computer time. Additional information concerning tunnel Mach number determination is contained in reference 9.

The conditions at the inlet throat were determined by computing the momentum and total enthalpy from the pressure forces and accounting for friction and heat losses incurred on the inlet spike and the internal surfaces. The inlet mass flow ratio and additive drag were determined from theoretical calculations (ref. 31). Pressures used in these calculations were obtained as follows: (1) for conditions where inlet start was obtained ($M_{th} > 1$), the calculated mass-momentum-average static pressure was used, and the measured static pressures at the throat were not used; and (2) for conditions where inlet unstart was experienced ($M_{th} \leq 1$), the average of the measured static pressures at the throat was used with the Mach number constrained to unity to calculate spillage and additive drag.

For both cases above, the flow was analytically expanded (isentropically) from the inlet throat conditions to the freestream static pressure in order to determine the hypothetical static enthalpy and associated velocity which are required to compute the inlet kinetic energy efficiency and the inlet process efficiency (as required under the contract statement of work). Also the flow was analytically compressed (isentropically) from the inlet throat conditions until the calculated total enthalpy matched the known total enthalpy after heat loss. For a started inlet, a side calculation was made by isentropically expanding the flow to an area which was arbitrarily set 10 percent larger than the throat area (for flow stability). At this point, the flow was passed through a normal shock. The limiting subsonic pressure recovery for the inlet and the corresponding kinetic energy and process efficiencies were then determined from conditions downstream of the normal shock. These inlet performance parameters were considered of interest as indicators of the overall inlet performance and of flow conditions prior to inlet unstart.

Two methods were used to calculate conditions at the combustor stations: (1) up to the first station where fuel was injected, the mass-momentum-averaged static pressure that satisfied the state, continuity, momentum, and energy equations was calculated; and (2) at stations downstream of the first fuel injector the average of the measured innerbody and outerbody pressures was used, and the combustor efficiency was calculated to satisfy the conservation equations. For these methods it was assumed that the flow area equals the geometric duct

area (no flow separation). The amount of hydrogen required to react in order to satisfy the measured static pressure, the duct area, the heat loss, and the conservation equations is computed by the program. Of the total hydrogen injected or present in the flow at a given station, the amount which reacts has been named "real" hydrogen and is used in the equilibrium chemistry process being completed. The hydrogen which is not reacting has been named "inert" hydrogen. The concept of real and inert hydrogen and the station-wise conversion from inert to real is simply a bookkeeping procedure in the program which simulates or "models" the mixing process. The inert hydrogen is assumed to have the properties of an inert gas, not to react with other species, and not to dissociate.

The combustor throat was defined as the point of minimum-flow area between the struts in the subsonic combustion mode and at the strut exit plane in the supersonic combustion mode. When the computed one-dimensional Mach number at the assumed combustor exit was found to be less than 0.95, the computation was considered to improperly represent the subsonic combustor flow situation in that the flow must have reached a sonic point further downstream. With the area increasing added combustion (heat release) downstream of the assumed combustor exit station is implied. Therefore, a side calculation was made of the combustor efficiency required to produce sonic velocity at the assumed combustor exit station, as if this added heat release occurred prior to the assumed combustor exit station. For this condition, the performance program printout shows results under the heading SONIC THROAT (e.g., reading 94, time 150.342 sec).

The regeneratively cooled combustor performance ("COMBUSTOR REGEN" in the performance program printout) was simulated by recalculating the total enthalpy at the combustor exit as the sum of the free-stream enthalpy of the synthetic air, the enthalpy of the hydrogen fuel at 50° R, and the absolute value of the heat loss through all the engine surfaces wetted by the internal flow stream. Using this total enthalpy, the stream total pressure, and the same combustion efficiency, the combustor exit static-state properties were also computed.

Nozzle performance was obtained by isentropically expanding the flow from the actual and regeneratively cooled combustor exits to the nozzle exit area and to ambient pressure ("NOZZLE AE" and "NOZZLE PO" in the performance program printout). The flow was then isentropically expanded from the actual combustor throat to those nozzle stations representing the locations of pressure taps, and the local skin-friction coefficients were calculated using the Spalding-Chi correlation. The nozzle vacuum stream thrust coefficient was also computed. This coefficient is arbitrarily defined in previous HRE documents (e.g., refs. 3 and 15) as the ratio of the actual nozzle exit total momentum (stream thrust) divided by the theoretical nozzle exit total momentum where the flow was isentropically expanded from the combustor exit conditions to the nozzle exit area (512.389 in^2). The actual nozzle exit total momentum was determined by taking the combustor exit total momentum and adding (or subtracting) the pressure force, the friction force, and one-half of the calculated drag force (onehalf of strut assumed to be charged to the nozzle component). The hypothetical static enthalpy resulting from the computed isentropic expansion from the combustor exit conditions to the free-stream static pressure was used to calculate the nozzle kinetic energy and process efficiencies.

Side calculations were made of a fictitious stagnation combustion process (constant pressure and zero velocity) with 100 percent combustion efficiency and no loss to the walls (denoted in the performance program printout as "FICTIVE COMBUSTOR"), followed by an isentropic expansion to ambient pressure to obtain the combustor effectiveness. Also to obtain the combustor effectiveness, the flow at the combustor exit was expanded to free-stream static pressure and the total momentum at this pressure was determined. The combustor effectiveness (ref. 15) is then the change in total momentum for the actual combustor process from the combustor entrance condition to the expanded (free-stream static pressure) condition divided by the change in total momentum for the fictitious process mentioned above from the combustor entrance condition to the expanded (free-stream static pressure) condition. Side calculations were also made of a fictitious nozzle to determine the static and total conditions ("FICTIVE NOZZLE" in the performance program printout) required to match the actual vacuum specific impulse at the nozzle exit.

Calculation of cooling load distribution. For the AIM tests, the heat loss distribution was determined from the differences between the skin thermocouples inbedded in the engine surfaces and the cooling water temperatures. Standard heat-transfer equations were used to obtain local heat losses. These losses were then adjusted linearly with the overall heat loss as measured by the overall water temperature rise. The detailed equations and procedures used for these computations are presented in reference 9.

Tare forces.- Purge nitrogen was injected in the AIM cavity between the non-metric "windshield" shroud and the metric outerbody to assure that hot tunnel gases did not enter into this cavity. This method produced a large tare force which was of the same order of magnitude as the engine net thrust. An effort was made to reduce and even control the tare force by suitable control of the pressures in two parts of the cavity. This tare-force control concept was, however, not achieved. Since the thrust is considered the most important measurement in evaluating the engine performance, special tare-force calibration tests were made and the results carefully correlated in order to determine the correction for the measured thrust. The method and procedures are described in detail in references 5 and 9.

External drag. - The external drag was calculated from the summation of pressure and friction forces acting on the external metric surfaces of the AIM. The method and procedures are described in reference 9.

Strut force calculation. The performance program was originally programmed to calculate strut force based on a theoretical calculation, assuming uniform flow ahead of the strut. This force should be a drag term since, theoretically, pressures downstream of the maximum strut blockage should be lower than upstream. However, test data indicate that this is only true with subsonic combustion. Upon examination of the test data, it appeared that measured static pressures between struts on both the inner and outer walls (there were no measurements along the strut surfaces) could be used to represent the forces occurring on the strut surface. Thus, a pressure integral was used to determine the strut force and a calculation was also made for strut base pressure as discussed in reference 9.

Performance correction for regeneratively cooled system. The AIM incorporated a water-cooled jacket in which heat was rejected and not recovered. In order to compensate for this heat loss, hydrogen fuel was heated up to 1500° R to simulate a regeneratively cooled system. The deficiency of energy in the system in terms of theoretical energy release was less than 10 percent in all cases.

In order to correct this deficiency, the performance computer program (ref. 6) incorporated a side calculation in which the energy deficiency, because of the heat loss through internal surfaces, was added to the stream at the combustor exit with no total pressure change. The flow was then expanded to the nozzle exit with measured nozzle efficiency. The differences between the heat added to fuel and the internal cooling loss are presented for several tests in reference 9 as table 6.6-1.

Performance correction for inlet total temperature.— Because of the facility heater deterioration, the true temperature simulation of 3700° R at Mach 7 was not achieved (the test Mach number was generally about 7.25 requiring a simulation temperature of about 3960° R). It is known that the effect of decreasing total temperature is to increase the engine performance. Therefore, it is necessary to correct the measured performance for Mach 7 (results contained herein) to properly account for deviations in test conditions. Theoretical calculations indicate that, at Mach 7, a decrease of 560° R would increase the thrust coefficient by 5 percent and the specific impulse by 3.5 percent. The accomplishment of this correction in the performance computer program (ref. 6) employed the methods discussed in reference 9.

Determination of tunnel gas composition.— The oxygen-to-nitrogen ratio was determined from the flow measurements of oxygen, diluent nitrogen, and nitrogen entering the storage heater, and checked by gas samples taken through two aspirating thermocouple probes 180° apart in the facility nozzle entrance prior to each run. The samples were collected in high-pressure bottles and later analyzed on a mass-spectrometer. The measured compositions for each run are presented in reference 9 as table 6.8-1. The one-dimensional performance computer program (ref. 6) used only the N₂ and O₂ values.

RESULTS

Selected points of interest of the HRE/AIM test data have been analyzed by use of the one-dimensional performance analysis computer program (ref. 6). The amount of material generated requires four volumes. Mach 7 component integration and engine performance results are presented herein. Mach 6 component integration results, Mach 6 engine performance results, and Mach 5 component integration and engine performance results are presented in references 10 to 12, respectively. All of these results were used in references 7 through 9 in the discussion of the results of the AIM test program.

Selected Test Points for Performance Analysis

Details of the AIM tests were discussed in reference 5 which included a list of all the HRE/AIM tests; this list is contained herein as table 1 (included in each volume). The individual AIM tests were recorded as consecutive reading numbers that extended through number 97 for a total operation time of 112 minutes with 41.5 minutes of combustor operations. About 60 successful tests are noted in the first column of table 2.

Reference 5 documented the fuel injection schedules, both planned and measured, for the successful tests. The measured fuel injection schedules for the successful Mach 7 tests are contained herein for convenience in figure 4. Such plots were reviewed and points (run time) of interest were selected for performance analysis. The selected points were listed in reference 9 and are included in tables 5(a) through 5(d) for the results presented in references 10 and 11, herein, and reference 12, respectively, where the times correspond to the abscissa in figure 4. The first column of table 5 indicates the page number of the initial page of the data for a given test point (specific time of a reading number). Table 5 indicates the general test conditions and fuel injection equivalence ratios, ϕ , for the first-, second-, and third stage injectors and the accumulative ϕ -value. Also, the use of ignitors is indicated and the general purpose of the test is noted.

Vagaries in the test program that should be noted (table 5, last column) are:

- (1) Fuel equivalence ratio values, ϕ , in table 5 for reading 93 are lower than the values indicated by the fuel injection schedule (fig. 4(a) of ref. 12). In preparation for the performance analysis, the tunnel measured oxygen content was found to be about 34 percent instead of the standard 21 percent; therefore, the fuel equivalence ratios were corrected to account for the difference in the available oxygen for combustion.
- (2) Time 235 seconds in reading 90 is for an inlet unstart condition. With an unstart, the captured mass flow is, of course, greatly decreased, and since the fuel flow rate is still high, the ϕ -value would be high as indicated, therefore this time is not very meaningful.
- (3) At Mach 7 the agreement between computed thrust (a function of ∫pda) and measured thrust was not nearly as favorable as experienced for Mach 6. Examination of the surface static pressure distributions on the outer combustor surface in the vicinity of the pressure rise indicated some pressure instrumentation to be faulty. For reading 89, more reasonable values were substituted for the measured pressures and the performance recomputed. The recomputation was performed for two different times, 316.47 and 327.27 seconds (see table 5(c)), and the results indicate a much more favorable agreement between the computed and measured thrust. The channel numbers in which new pressure values were substituted are noted on the first page of the results for these two times. A more detailed discussion of this exercise is contained in reference 9 (section 7.7.2 Mach 7 Performance).

- (4) Times 264.04, 274.84, and 275.74 seconds of reading 96 had a fuel flow measurement malfunction that indicated no fuel flow from injector 1B. Injector 1B manifold pressure, however, indicated flow to exist at pressure levels about equal to planned pressure levels (ϕ -values about the same as for injector 1A). The performance calculations for these times of reading 96 erroneously used only fuel flow from injector 1A.
- (5) At time 313.54 seconds, also of reading 96, the test chamber pressure was noted to be high, thus yielding unrealistically high pressures on the AIM nozzle shroud and plug that would, of course, contribute erroneously to increased engine thrust.

Description of Performance Computer Results

The selected points listed in table 5 were analyzed using the performance computer program described in reference 6. As noted in the Method of Computation section, the AIM test data were reduced to engineering units and reviewed for erroneous data. Such data were "coded out" in the performance computer program. Table 6 indicates the channels that were coded out. The COXX indicates the code outs for a reading number, e.g., for reading 33, CO33 is indicated. Channels that are coded out are listed adjacent to the notation KODSEL, e.g., for reading 33 the first and last of 85 coded out channels are 60 and 399, respectively. The locations and type of measurement for the listed channels may be determined by referring to table 4.

Several points (run time) of interest were selected for each run as indicated in table 5. The page numbers indicated in the first column of table 5 are output listings of the performance computer program (ref. 6). For each time of interest there are seven or eight pages of computer output listings. On each of these pages a standard heading exists: READING number (test number); BLOCK number (numbered sequentially and corresponding to recording times of test data); TIME (of data recording, seconds); MACH number (in wind tunnel); PT (total pressure in wind tunnel, psia); TT (total temperature in wind tunnel, OR); and PAGE number.

Station flow parameters.— A summary of flow parameters at each calculation station in the AIM is contained on pages 1, 2, and 3. Each station is headed by a station designator (i.e., WIND TUNNEL, INLET THROAT, COMBUSTOR, etc.), followed by three integers (the zero following the combustor designator is meaningless). The first integer denotes the station number, the second denotes the combustor station, and the third denotes the number of interations required to converge on a solution. The third integer may assume values between 0-21, 100-121, and 200-221. A value of the third integer equal to 21 denotes that the mass flow was too great or the flow area too small to obtain a solution, 121 denotes that the solution for total conditions did not converge in 21 interations and 200-221 denotes that the mass flow was too small or the flow area too large to obtain a solution. When both solutions for static and total conditions have converged, the third integer may assume the values 1-20 or 101-120 depending upon which solution (static or total) required the larger number of interations. Columns 2-8 have two rows of values for each station; total and static conditions in first and second rows, respectively.

Most of the station designators are self-explanatory. The first appearance of the designators WIND TUNNEL and SPIKE TIP NS (NS = NORMAL SHOCK) reports conditions in wind tunnel and upstream of the spike tip based on a wind tunnel Mach number determined from calibration runs. The second appearance of these designators reports these conditions based on a wind-tunnel Mach number calculated from the total and pitot pressures and the total temperature of the synthetic air applied to the normal shock equations. The designators INLET UPNRSK and INLET DNNRSK denote conditions upstream and downstream of a normal shock positioned at a fictitious flow area 1.10 times the flow area at the inlet throat. The designator COMBUSTOR REGEN denotes, for cases with fuel flow, conditions at the combustor throat simulating a regeneratively cooled ramjet. In some cases (e.g., reading 94 time 150.342 sec) the designator SONIC THROAT appears ahead of the COMBUSTOR REGEN. This denotes the results discussed in section entitled "Description of Performance Program Methods." NOZZLE AE and NOZZLE PO report conditions when the flow is expanded isentropically to the nozzle exit area and to the windtunnel static pressure, respectively. NOZZLE AE REGEN and NOZZLE PO REGEN denote, for cases with fuel flow, conditions at the nozzle exit simulating a regeneratively cooled ramjet. FICTIVE COMBUSTOR denotes stagnation combustion conditions (zero velocity and constant pressure) with combustor efficiency equal to unity. FICTIVE NOZZLE reports conditions required to match the actual momentum and nozzle exit

Definition and units of parameters in the SUMMARY REPORT, pages 1-3 in the computer listings, are listed below:

P - pressure, psia T - temperature, OR H - enthalpy*, Btu/lbm GAMMA - specific heat ratio MOLWT - molecular weight SONV - conic velocity, ft/sec MACH - Mach number VEL - flow velocity, ft/sec S - entropy, Btu/lbm-OR W/A - flow rate per unit area, lb_m/sq in W - flow rate, lb_m/sec A/AC - mass flow ratio MØMTM - flow momentum, lb_f Q - dynamic pressure, lb_f/sq in IVAC - vacuum specific impulse, lb_f-sec/lb_m PHI - equivalence ratio (see discussion in Ramjet Performance section) ETAC - combustor efficiency

where: $C_{p,j}$ is specific heat at constant pressure, Btu/lb_m - OR, and $\sigma_i(T)$ is the mass fraction of the specie i as a function of temperature and H_f is fuel enthalpy. 12

^{*}Two values were reported. The first value (column 4) was the JANNAF-based enthalpy. The value in parentheses (column 5) was the enthalpy potential or the sensible enthalpy based on the equation

Cooling and surface-pressure parameters. - Surface pressures, cumulative surface-pressure integrals, cumulative cooling losses, cumulative surface area, and pressure ratios for axial distances from the AIM virtual spike tip are listed on pages 4 and 5.

Definitions and units of the parameters are as follows:

```
XABS - axial distance from virtual spike tip, in P-IB - surface pressure on innerbody, psia P-ØB - pressure on cowl inner surface, psia PDA - cumulative surface-pressure integral, \int_{0}^{XABS}_{PdA}, lb_{f}^{DdA}

QØX - cumulative total cooling loss, Btu/sec Q-IB - cumulative cooling loss from innerbody, Btu/sec Q-ØB - cumulative cooling loss from outerbody, Btu/sec CAWALL - cumulative surface area, sq in P-IB/PSØ - innerbody static to wind-tunnel static-pressure ratio P-IB/PTØ - innerbody static to wind-tunnel total-pressure ratio POB/PSØ - outerbody surface static to wind-tunnel static-pressure ratio POB/PTØ - outerbody surface static to wind-tunnel total-pressure ratio
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Drag and heat-transfer coefficients. - Longitudinal values of drag force and drag and heat-transfer coefficients are listed on page 6 (for some cases on page 6 and 7). Definition and units of the parameters are as follows:

```
X - axial distance from spike virtual tip, in DDRAG - incremental frictional drag force, lf_f CDRAG - cumulative frictional drag force, lf_f Cp - friction-drag coefficient HC - heat-transfer coefficient, Btu/(sec-sq\ ft^{-O}R)
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Ramjet performance. AIM performance parameters and pertinent information are contained on page 7 (page 8 for some cases). The performance parameters are generally self-explanatory; detailed discussion about the methods of computation are presented in references 6 and 9. Parameters listed below STATIONS are presented since they are related (except for the inlet throat) to the cowl leading-edge station. The NOMINAL COWL LEADING EDGE refers to the χ_{CL} (table 3) value for the Mach 6 design operating position. SPIKE TRANSLATION is the recorded distance between the nominal and the actual χ_{CL} value (this distance is designated as $\Delta\Delta x$ in symbols and used in figure 3(a)); all dimensions other than those for the inlet spike are corrected by this amount.

The fuel injectors and their corrected stations in inches are shown. A letter in the VALVE column indicates the injectors that were in use during the respective time. Table 5 indicates the general fuel equivalence ratio values for the various injector stages. The actual fuel equivalence ratio, however, for each injector can be determined by noting the step increases in the PHI column on the output, pages 1-3, for the respective time (ignore 0.01 or 0.02 changes); the step difference at the combustor station corresponding to the indicated injector station is the $\phi\text{-}\text{value}$ for the respective injector.



SUMMARY OF TESTS

The Hypersonic Research Engine/Aerothermodynamic Integration Model was tested in the NASA Hypersonic Tunnel Facility at the Plum Brook Station of the NASA Lewis Research Center. Synthetic air (heated nitrogen with proper amount of oxygen added) was delivered by the facility at nominal Mach numbers of 5, 6, and 7. The Mach 5 and 6 tests were conducted at true air temperature while Mach 7 tests were conducted at Mach 6 temperature (3000° R) because of heater deficiency. Changes in total temperature and instream oxygen content at Mach 5 and 7 were also explored. The hydrogen fuel was heated up to 1500° R prior to injection to simulate a regeneratively cooled system.

The engine testing was completed with an accumulated actual running time of about 112 minutes with 41.5 minutes of combustor operation. The important achievements realized from this test program which advanced the state-of-the-art in hypersonic propulsion were discussed in detail in reference 9 and are:

 Realistic engine performance levels for hypersonic flight were obtained from Mach 5 to 7.

Test Mach No.	Equivalence Ratio	Internal Thrust Coefficient	Internal Specific Impulse
5.1	1.0	0.910	2740
6.0	1.0	0.735	2360
7.25	1.0	0.570	2170

- 2. Engine inlet performance agreed well with theoretical prediction. Combustor efficiency of 95 percent was achieved. Nozzle vacuum thrust coefficient was lower than predicted.
- 3. The interaction effects in staged fuel injection were very important in achieving auto-ignition, high combustor efficiency, and overall performance. High supersonic combustor efficiency in a diverging duct was difficult to achieve. The strong stage interaction effects discovered during these tests may be used to great advantage in future designs.
- 4. The "transonic combustion" or "mixed combustion mode" was the most efficient heat addition process in the range of Mach numbers and temperatures tested in this program.
- The effects of ignitors, altitudes, spike translation, fuel schedules, angle of attack, step and struts, inlet gas composition, inlet total temperature, and component interactions were investigated and correlated.

- 6. Stable subsonic and supersonic combustion and convertibility over a range of fuel equivalence ratios at Mach 5 and 6 was demonstrated.
- 7. The overall cooling load and its distribution as compared with theoretical prediction was determined.
- 8. Experience was acquired in free jet testing in a ground test facility with large model blockage and combustion.

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Table 1. - Summary of planned HRE/AIM wind tunnel tests.

(obtained from ref. 9 and 15)

RUN	M ₀	PTO, PS IA	TTO, °R	~	FUEL - SYSTEMS	FUEL SCHED.	INLET	COMBUSTION MODE	RUN TYPE AND PURPOSE
,	٥	466	1500	٥	•	-	4.23	-	Purge force, nominal case
2	6	466	1500	0	-	-	1.90	} -	Purge force, effect of spike position
3	6	466	1500	3	•	-	-4.23	-	Purge force, effect of angle of attack
4	ا ه	466	2000	0	-	-	4.23		Operation checkout, effect of higher TTO
5	٥	466 -	3000	0	-	٠	0, 1.71, 2,52 4.23, aft stop	-	Airflow calibration, effect of altitude
6	6	930	2946	٥	•	-	0, 1.71, 2.52 4.23, aft stop	-	Airflow calibration, nominal case
7	¢	930	2946	3	-	•.	0, 1,71, 2,52 4,23, aft stop	-	Airflow calibration, effect of angle of attack
6	6	930	2946	٥	14. 1b	' '	4.23	Supersonic	Inlet-combustor performance, ignition and inlet unstart limits
9	٥	930	2946	0	la, 16. 2a, 2c	2	4.23	Supersonic	Inlet-combustor performance, injector optimization
10	٥	930	2946	0	Ic. 4, 2a, 2c	2	4.23	Supersonic	Inlet-combustor performance, injector optimization
-11	٥	930	2946	٥	ia, lb, ic, 6	3	4.23	Supersonic	Inlet-combustor performance. Injector optimization
12	٥	930	2946	٥	TBD	TBD	4.23	Supersonic	Inlet-combustor performance, injector optimization
13	6	466	3000	0	la, 16, 2a, 2c	2	4.23	Supersonic	Inlet-combustor performance, effect of altitude
14	6	700	3000	0	la. 16, 24, 2c	2	4.23	Supersonic	Inlet-combustor performance, effect of altitude
15	6	930	2946	0	la, lb, 2a, 2c	2	Aft stop	Supersonic	Inlat-combustor performance, effect of spike position
16	6	930	2946	0	ia. 15, 2e, 2c	2	2.52	Supersonic	Inlet-combustor performance, effect of spike position
17	6	930	2946	0	1a, 1b, 2a, 2c	2	1,71	Supersonic	Inlet-combustor performance, effect of spike position
18	6	930	2946	0	3e. 3b	4	4.23	Subsonic	Inlet-combustor performance, subsonic combustion
19	٥	930	2946	0	3a. 3b	5	4.23	Subsonic & transition	Engine performance, subsonic combustion and transition
20	٥	930	2946	0	la, lb, 2a, 2c	2	4,23	Supersonic	Engine performance, nominalicase
21	٥	466	2946	0	la, 16, 2a, 2c	Z	4.23	Supersonic	Engine performance, effect of altitude
22	6	930	2946	3	le, 15, 2e, 2c	2	4.23	Supersonic	Engine performance, effect of angle of attack
23	,	520	1500	0	_	- 1	₹.88		Purge force
24	7	520	3965	٥	- .	-	2.34, 2.88 3.24		Airflow calibration, effect of altitude
25	7	1000	3840	0	-	•	1.98, 2.88 3.24	•	Airflow calibration, nominal case
26	7	1000	3840	3	-	-	2.34, 2.88 3.24	-	Airflow calibration, effect of angle of attack
21	7	520 A	3965 3840	0	le, lb	6	2.66	Supersonic	Inlet-combustor performance, ignition and inlet unstart limits
28	7	1000	3840	0	le, 16, 2e, 2c	,	2.88	Supersonic	Inlet-combustor performance, injector optimization
29	7	1000	3840)°	lc, 4. 2a. 2c	'	2.88	Supersonic	Inlet-combustor performance, injector optimization
30	7	1000	3840	l°	la, 1b, 1c, 4	8	5.66	Supersonic	Inlet-combustor performance, injector optimization
31	7	1000	3840	0	TBD	TBD	2.68	Supersonic	Inlet-combustor performance, injector optimization
32	7	522	3965	°	!a, Ib, 2a, 2c	7	2.88	Supersonic	Inlet-combustor performance, effect of altitude
33	7	700	3965	١°	la, 15, 2a, 2c	7	2.88	Supersonic	Inlet-combustor performance, effect of altitude
34	7	1000	3840	0	la, 1b, 2a, 2c	7	3.24	Supersonic	Inlet-combustor performance, effect of spike position
35	7	1000	3840	0	la, lb. 2a, 2c	,	2.34	Supersonic	Inlet-combustor performance, effect of spike position
36	7	1000	3840	0	la, lb, 2a, 2c	7	1.98	Supersonic	Inlet-combustar performance, effect of spike position
37	7	1000	3840	١°	la, lb, 2a, 2c	7	2.88	Supersonic	Engine performance, nominal case
38	7	522	3965	٥	la, 1b. 2a, 2c	7	2.88	Supersonic	Engine performance, effect of altitude
39	7	1000	3840	3	ia, ib, 2a, 2c	7	2.88	Supersonic	Engine performance, effect of angle of attack
40	5	445	1500	٥	la, lb, 2a, 2c	-	4.23	•	Purge force
41	5	206	1	0	la, Ib, 2a, 2c	-	4.23	•	Airflow calibration
42	\$.	415	2210		la, 16, 2a, 2c	9	4.23	Supersonic	
43	5	415		0	la, 16, 20, 2c	TB0	4.23	Supersonic	Inlet-combustor performence, and ignitor flow rate
44	5	415	2210	0	1a, 1b, 2a, 2c	9	4.23	Supersonic	
45	5	415	2210	0	3a, 3b	10	4.23	Subsonic	Engine performence, subsonic combustion
46	5	415	2210	3	la, ib, 2a, 2c	"	4.23	Subsonic & Supersonic	Engine performance, effect of angle of attack

Table 2. - HRE/AIM Test Run Summary (obtained from ref. 5).

												_	_													
		Comments	Deta not valid due to mechanical interference between AIN and outer cost body	Tast terminated due to cooling system overpressure abort system	Tunnel nozzle started. Inlet started. Strong shocks in test section. Cell pressure ~ 2.0 psie.	Test aborted due to facility problem (IAFP).	Facility shroud extended and washer added to assist tunnel start (TAFP).	TAFP	Mozzla start and inlet start obtained Cell pressure - 1.2 psis, whose mozzle pressure changed from 20 to 60 psis. No Improvement in eall pressure.	TAFP	. Wedge nozzle pressure 55 to 90 psig. No tunnel nozzle start. Nozzle started when inlet closed for shutdown.	TAFP	UAFP	First combustion attempt, TAFP	Nozzle start not obtained, TAFP.	Mazie start obseined by cycling inles spike open and closed, inles start obseinade. Fuel ramped to equivalence rasio = .35 prior to tunnel unstart and TAFP.	Mazzle stert with injet pertially open. (Δ x = 0.99). TAFP. No fuel injected.	No start at $\Delta t = 0.99$. Nozzle started by cycling inlet spike, Combustor lit causing tunnel onstart.	Jet pump installed. Test aborted due to freezing of coolant supply system.	Jet pump used for this test. Mozzle start obtained. Unstart asperienced when inter was sperienced when inter was sperienced annually. Restair essent nated during shutdom.	Jet pump and wedge nozzle inlet pressure veried, Mozzle start was not obtelned. Use of jet pump did not affect test chamber pressure. Seals between Alis support struts and facility shroud bloam out.	Jat pump inactivated, TAFP	TAFP	Mozzle start and engine start obtained. Fuel injected for by seconds prior to nozzle unstater. Unstare attributed to consister fuel injected caused by facility walve mateurion.	Mozzle start and inlet stort obtained. Jet pump inactivated, fun was injected, engine inlet unstart experienced 12 seconds later. Inlets start restablished and fun again injected inlet unstart experienced 9 seconds later. Test was manually aborted, Coul leading edge assembly spearated from the outer body. Gause of the separation was attributed to failure of the screw heads. The failure was caused by overheating of the intention of the start heads resulting from ingesting the hot tunnel environment into this area. Ingestion of funnel ambient was the regult of Additional idagons it instrumentation was installed in the facility shroud and diffuser.	Tunnel configuration same as config. B except washer inside dismeter changed to 44,5; inches. Tunnel unstart observed 19 seconds after fuel introduced, Start restablished, Test manually aborted 3 seconds later when excessive heating of MECA-11 coel leading edge assembly mount flange was noted. Excessive heating of the external skin of the All was noted.
		Objective of Test	Pre-run reference No-airflow engine Purge system calibration	Facility and engine checkput	Same as run 2	Establish facility operational procedure	Same as run 4	Same as run à	Same as run to	Seme as run &	Seme as run &	Same as run &	Same as run 4	Same as run &	Same as run &	Some as run 4	Same as run 4	Same as run 4	Establish facility operational procedure	Same as run 12 above	Same as run 12 above	Same as run 12 above	Same as run 12 above	Same as run 12 above	Same as run 12 above	Establish facility operational procedure to obtain hypersonic airflow.
П	5	ž	,		-	-			8								·	,								
	Useful	Ę	,	•	•	•		·	·	•	•		·	·	•	•	٠	•			•	,	1.		•	
1 2	٠	Sec	•	ş	92	5		٠	39	•	. 40	ž	91	35	8	8	13	20	•	22	3 3.	•	•	6,7	3	= .
	ā	E .	_	•	7	•	-	-	2	٠	-	•	•	·	-	•	•	-	'	•	•	1	'	'	-	-
*	Tunnel	Conf 19.	*	٧	₹	۷	18			18		18		-			<u>-</u>	10	5	ច	5	23	2	2	8	95
و	Injectors	Used			•	•	•	•		•	•	•		1C. 4		10, 4		16, 4	•		•	•	•	1A, 18		1A, 18
inlet	Position,	∆x, in,	•	4. 266	4. 266	4. 266	4. 266	4. 266	4. 266	3.962	3.962	3.962	4. 266	4 . 266		•	0.99/ 4.00	6.99/ 4.88	0.99/ 4.00	00°7 766°0	0.99/	0.99/	06.0 766.0	0.99/ 4.00	6.99/	0.99/ 4.00
8	•	10.	•	1500/2100	1500	1500	1500	1500	0051	2250	2250	2950	2950	2800		•	2950	2950	3000	3000	3000	3000	3000	000€	3100	3100
Inlet Condition	ة	10.	į.	994	994	799	994	9947	994	994	994	994	994	994			997	994	95/	750	750	750	750	95/	οτε	930
-	Rech	ė		9	6	9	6	9	•	6	9	9	6	٥	-		6	9	6	٠	۰	9	٠	9	10	•
	<u> </u>	-+	3/14/12	10/31/72	1/11	11/2	91/11	11/16	91/11	11/31	11/51	11/21	12/8/21	1/18/73	-	•	1/1	1/1	8//5/13	12/2	12/2	1/33	1/23	82/2	1,4	3/16
	Reading	ŝ.	f through S	,	,	80	9	.01	=	12.5		4	15	. 91	12	e e	61	02	≅	z	æ	24	25	3 ¢		98
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* see figure 5-9, reference 5

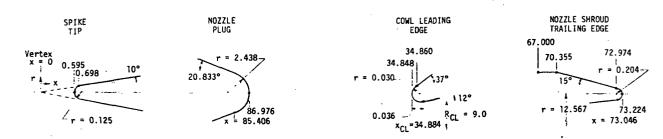
Table 2. - Continued.

	Comments	Re-run of reading 23 with seal repaired. Jet pump did not improve tunnal start,	Shroud inlet washer replaced with cone-cylinder and 190 conical diffuser inlet contraction replaced with 70 cone; tunnel nozzle did not stert.	First un with fully storted tunnel. Shroud intet come cylinder replaced with original 46 in. diameter washer. Tunnel start obtained when inlet spike was cycled twice; supersonic flow in diffuser. Test terminated when terget conditions achieved due to liameted supply of nitrogen. Test call pressure was 1.2 psie.			furmel start and inlet start obtained, d of 1.35 set at Pro = 150 pais and d of 1.00 set at Pro = 350 pais. Facility fuel control value for injector 18 ostillated. Nun proved AlM and tunnel can operate at d > 1.0. Frosion of zirconium oxide coasting on outer coal body crossover manifold noted. Erosion caused by carbon dust in tunnel flow.	Test was aborted when engine inlet unstart was observed three seconds after intitiation of feel injection. The angine unstart was sealt of injection accessive fuel, caused by malfunction of facility control valve. Inspection of the unit revealed that the coolent leak on the spiles assembly had progressed, and repair was nocessery.	First good run with design injector locations. Auto ignition obtained at g = 0.55; first stage did not light until second stage fuel added. Overall g ramped to 1.0 with first stage g hald at 0.28.	Test aborted due to maifunction of the steam ejector system	Test aborted when Inlet unstarted. Melfunction of the facility fuel control valve meuited in injecting axcessive fuel into injector 20, 3 smell cracks in spike skin in region of ignitors found in post run inspection. Cracks repaired to prevent water leak into combustor.		TAFP	Fuel control problems encountered.	Investigating performance improvement due to injecting fuel closer to iniet. Iniet unsterted at overall & of .83.		Attempt to determine effect of first stage g and thrust on performance. Auto ignition obtained at g = .54. Date taken with ignitors an and off to determine effect on performance. Inspection of unit revealed accessive coolant leak at spikel ignitor body interface. Repair necessary. Tunnel operating procedure modified to reduce water ingestion into AlM wall pressure teps.		Effect of fuel split between 1st and second stage injectors at overall 6 = 10 investigated. Also all second stage fuel added from innerbody side (system 22). Fuel system purges turned off to determine effect on combustor well pressure distribution. Found thrust measurement effected by thermal aspainton of fuel manifold IB. Inlet unstarted at overall 6 of 10 with first stage 8 = 0.36. Cavity pressure tep PAZ repaired for this run. Encountered fuel control problems.
	Objective of Test.	Same as run 17 above	Same as run 17 above	Establish operational procedure	Determine effect of varying wedge nozzle flow	investigate inlat unstart limit with first stage combustion	Checkout Aim and facility, Fuel rich at Pro = 750 psie g= 1.0 at Pro = 930 psie	checkout AIM and facility, Dasign.in- jector locations	Demonstrate operation with design injector location and datermina auto ignition limit	Determine effect of first stage # on com- bustor performance	Dotermine effect of first stage 6 on com- bustor performance	Purge system calibration test	Combustor optimization	Combustor optimization	Combustor optimization	Purge system calibration test. Evacuated test cell.	Combustor optimization	Purge system calibration evecuated test call	Combustor optimization
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		Objective of Test		Purge system calibration Determine affect of thermal expansion of fuel manifold 18,	Combustor optimization TAFP	Combustor optimization Overall g hald constant while amount of fuel from Innerbody and outerbody lightcore varies. Fuel temperature community pensation added to fuel control	Determine affect of inlet massflow ratios of 0.81 and 0.58 ran by varying the inlet spike position on spike position. All wall pressure distribution measured with amplied section of the innerbody standard and damaged during combustion; damaged section mass removed. Operational procedure modified to prevent further damaged.	Performance test TAFP	Performance test Tunnel total pressure varied to determine effect of altitude on performance.	Subsoric-supersonic Transition from subsonic to supersonic combustion mode demonstrated. Inspection of unit revealed coolents was floaring transition. Into the 18 fuel manifold and a nickel plated section of the innerbook hed blistered. Separation of the stimulation belongs to provide the spirit stimulation of the provided to approximately 1.0 inches. Tomand facing step at the interface of the coal leading object search and the pour problem of the coal leading object search to the part of the	Supersonic combustion instrumentation rake installed. Rake caused tunnel to unstart with instrumentation rig et e 1.05. Exhaust gas sampling data sekan,		Purge system celibration Mg purge force celibration with cell evecuated.	Time of steady state fuel flow increased to 20 seconds to allow yes sampling date to stabilize.	Supersonic combustion One tunnel unstart experienced near and of run. Several tunnel unstarts prevented by shutting off fuel. Incipient unstart detected by wonitoring luminescent normal shock position in T.V. view of tunnel.	Determine effects of Test terminated prematurally due to frozen vent valve, angle of etteck	Com! leading adge assembly removed efter this run to remove facing step noted after reading 66.	Aurge system celibration Celibration with 18 fuel Injector manifold heated test cell everaled.	Mach 7 facility chack- Test aborted due to facility problems (TAFP)	Mach 7 facility check- TAFP	Mach 7 facility check- Attempt to start tunnal at Mech 7 unsuccessful. Secondary staem out aleston to sad, medge nozzle pressure varied; inlet spike assembly translated.	Mach 7 facility check- Test aborted while attempting tunnel start, TAFP, Unusual out	Facility chack-out AIM moved aft 5.5 inches.	Facility check-out 1AFP (donar water system frazen). Facility check-out Blowout doors installed in tunnel closure. Tunnel started when	reuge nozze pressure reduced. Junel unitated when combustor I'i. Restart not obtained due to change in wedge nozzle inlet pressure.		Facility check-out Tunnel start not obtained.	
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		_		,								
			First successful Mach 7 run. Tunnel closure removed. Diffuser seal respirate, Effect of fuel injection location Investigated, where I spifors on, Outer coal body support damaged by carbon particles in tunnel flow due to failure of carbon part in facility hearist. Shroud inlet pressure rake hit and damaged. Repaired outer coal body support and water cooled protective wedge installed. Coolent lask at the Interface of piles shirt and sight body noted at anysist location 270° in addition to leak at 180 degress noted in Rag 64. Leak at 180° progressed to approximately 1.25 inches. Coal leading adderties remorted.	Performance measured with various fuel injection schemes. Tro waried during run. I upitors on. Test terminated prematurely due to failure of transducer in fuel control custing fuel control custing fuel control valve to fully open. Abnormal emount of carbon dust observed in turnel flow, Cowl loading edge tip radius and spike it p again demaged. Tip section repaired.	Second stage fuel injection closer to inlet (injectors 16, 4). Inlet unstarts encountered.	Tunnal start improved at angle of attack. Tunnel started at Pnn #80 psia, 3 inlet unstarts ancountered due to excessive ist stage fuel. Total coolant leak into combustor assimated to be 5.0 gpm.	instrumentation rate blockage had adverse effect on turnel start, that strucked tales to start turnel. Oxygen content of turnel flow varied while Alk enhust gas sempling date talen.	First Mach 5 run, Subsonic combustion data obtained, Run terminated promaturely (TAFP).	Subsonic and supersonic combustion and transition demonstrated. Four unstarts appearance thereof, these unstarts attributed to high call pressure, one to injecting excessive fuel intentionally into the AIN. More carbon in tunnel flow. Coal leading edge and spike tip demaged. Both remorked.	All comments made for Rdg 94 applicable for this run, except tembustion was limited to supersonic consultation was limited to supersonic consultation was leading. Four engine unstairts asper lenced. Prese unstatts were attributed to facility conditions and the other to programmed to determine inlet unstart limit.	Subsonic and supersonic combustion and transition demonstrated at angle of etteck. Intentional engine unstant obtained when excassive fuel was injected in supersonic combustion mode.	Combustor exit flow conditions surveyed. Gas sampling data taken, Blockage of instrumentation rake had adverse effect on tunnel operation.
	Objective of Test	objective of the	Combustion evaluation	Combustor optimization	Combustor optimization	Effect of angle of attack	Combustor performance with instrumentation rake installed.	Facility chack-out	Combustor optimization	Combustor optimization	Evaluate affacts of angle of attack	Combustor performance with instrumentation rake installed
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- = =	un S	366	5.7	•	8	¤	2	8	8	3		
	•		~	_	_	2	~	•	~	_		
	Tunnel	.g. Luon	L .	•		L	L			L.	•	u.
9	Injectors	03.60	14, 18, 24, 20	14, 18, 24, 20, 4	1A, 18, 1C,	1A, 18.2C.	1A, 18,2C,	1A, 18, 2A, 34, 38	7 X X X X X X X X X X X X X X X X X X X	14, 18, 24, 26, 14, 18, 24, 20, 14, 19, 24, 20,	14, 18, 24, 34, 38	2A, 3A, 3B
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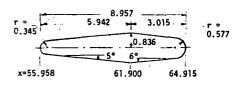
Table 3. - AIM aerodynamic coordinates (Mach 6 cowl position, $x_{CL} = 34.844$ in.)



a) Centerbody

	a, certer body
x, in.	r, in.
0.595	0.0 90°
0.698	0.123 st. line 3.237 10°
18.360	
19.304	3.411
20.443	3.633
21.691	3.885
22.830	4.122
23.850	4.338
25.875	4.782
26.766	4.985 i
27.900	5.256
28.904	5.518
29.655	5.726 \ 15.819°
30.360	
32.760	6.660
34.080	7.140 8.607) 22.0°
37.710	8.607/ 22.0
38.070	8.734
38.538	8.874
38.826	8.942
39.132	9.000
39.780	9.096
40.500	9.180- 5.645° Throat
42.000	9.318
43.400	9.415
44.000	9.452
45.000	9.518
46.000	9.578
47.000	9.624
47.600	9.650 9.670
48.400	9.670 End of
55.760	9.406 spike; step 09
55.760 61.900	9.406 Spike; Step 10-
65.740	9.406 (nerma) throat:
67.553	
85.406	9.072 2:278 20.8330
86.976	0.0 900
00.370	. 0.0 .0

c) Internal struts (6)



b) Outerbody

	b) odcerbody
x, in.	r, in.
40.894 36.750 36.250	11.611 10.103 9.975 9.808 External
36.000 35.750 35.437 34.860	9.685 9.487 9.053 37°
34.848	9.029 90°
34.884	9.000 12°
35.397	9.104
35.874	9.192 10°
36.171	9.241
36.414	9.278 8°
36.765	9.322
37.494	9.398
40.500	9.695) 5.645°
40.894	9.720
41.894	9.810
42.894	9.890
43.894	9.960
46.294	10.132
55.760	10.873
57.000	10.955
58.000	11.000 Internal
58.700	11.022
61.900	11.022 - Thermal
65.980	11.022 - throat
66.220	11.042
66.740	11.132
67.740	11.348
68.780	11.572
69.740	11.773
70.820	11.989
71.660	12.146
72.260	12.249
72.920	12.349
72.980	12.357
73.046	12.365
73.224	12.567 90°
72.974	12.791
70.355	13.493 } 15° External
67.000	13.493

(d) Cowl lip design positions

	xCL, in-	Δx , in	*CL/RCL
Close off	39.150	0.0	4.350
Inlet start	38,160	0.990	4.240
Mach 8	36.990	2.160	4.110
Mach 7	36, 270	2.880	4.030
Mach A - 6	34.884	4.266	3.876

Table 4. - HRE/AIM Instrumentation (obtained from ref. 5).

(a) Coding for instrumentation list.

The code for the instrumentation listed in the "Identification" column is as follows: Sample, $S-P-14.492-0^011!-90-3$ (A-B-C-D-E-F).

"A" designates the component on which the instrumentation is located:

\$ = inlet spike assembly

I = innerbody assembly

NP = nozzle plug assembly

CO = cowl leading edge assembly (outside)

C = cowl leading edge assembly (combustor side)

0 = outerbody

N = nozzle shroud (combustor side)

NO = nozzle shroud (outside)

CE = combustor exit

EF = engine airflow-metering duct

F = fluids

"B" designates type of instrumentation

P = pressure

T = temperature

"C" designates the location of the instrumentation in terms of station, with the inlet spike assembly positioned for testing at Mach 6 condition.

"D" designates the angular location in degrees and minutes.

"E" designates position of the pressure pickup with respect to airflow in degrees, or, if the instrument is a temperature sensor, it designates the thermocouple:

CA = chromel alumel

CuC = copper constantan

P/rh = platinum-platinum/rhodium

"F" designates the leg through which the leads are brought out.

An "X" anywhere in the Identification Code indicates that the parameter was not applicable.

xxx/yy in the "Reading No." column indicates the Channel No. (xx) on which the parameter was recorded, and the rated capacity (yy) of the transducer used.

The "N/U" Code in the "Reading No." Column indicates channels that were not used.

"LeRC Sys" - recorded on separate system, therefore no channel number.

Table 4. - Continued.

(b) Instrumentation list.

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Table 4. - Continued.

(b) Continued

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280 - 355 - x	172/50	172/15	172/50
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	177/50	177715	174/50
330 - 00 - x	1,18/75	178/50	178/75
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Table 5. - Summary of HRE/AIM test points used for analyses.

(a) Mach 6 component integration results:

Page No.	Reading Number	Time	M,	٥	P _T o	a	T _T o	· x	CL'	a	Inj.1/0 ₁	Inj.2/ø ₂	1nj.3/ø ₃	φ _T	ignitors	Purpose & Remarks
	33**	126.95	6.0	.	75	٦	3000	٦,	5.2	o	0	0	0	0	No	No fuel injection
	1	161.15		\vdash	Ť	+	Ť	+	Ť	Ť	1A,18/.24	0	0	0.24	1,2	1st stage only
		168.0	Н	H	一	+	+	十	T	H	1A,1B/.3	0	0	0.30	<u> </u>	1st stage only
	 	174.65	H	H	┪	+	+	+	╁	1	1A,18/.36	0	0	0.36	 	Max. Ø, engine unstart
	21.					7	2000	+	-	00	0	0			-	
57 65	34	98.15	6.0	ř	- 75 	4	3000	+-	5.2 T	H	1A,18/.20	0	0	0	. 1,2	
73		148.55	Н	H	\dashv	+	+	+	+	H	1A,18/.23		0 .	0.20	- -	1st stage only
81		181.85		H	1	+	+	+	╁╌	H	1A,18/.21		3A/.39	1.16	 	1st and 2nd stages Max. Ø, 3 stages
89		196.25	H	,		٦	╁	+	+	H	1A,18/.15		3A/.32	0.91	 	Max. Ø, 3 stages
						\exists		1	-							
	36_ a		6.0	-	75	"	3000	43	5.2	0	0	0	0	0	No	Auto ignition
	l a	124.58	\vdash	╁		+	+	+	+	╁	1A,1B/.26	0	0	0.26	\vdash	1
97	\vdash	132.68	\vdash	H	-+	+	+	+	╀	ℍ	1A,1B/.25		0	0.59		
106	 	144.38	\vdash	H	\dashv	+	+	╀	╁	₩	1A,1B/.24	2A,2C/.49	-	0.73	 	
115	┝╼╁╼╾	173.18	-	\vdash	1	+	\pm	+	╁	H	1A,1B/.23	2A,2C/.69		0.92	 	
	-		\vdash			#		#	-	Ľ	1A,1B/.22	2A,2C/.75		0.97		
133	38	96.24	6.0	1	75	<u>• </u>	3000	1 3	5.2	00	0	0	0	0	No	
141		107.05	Ш	\sqcup	4	4	4	+	↓_	Н	IA, IB/.33	0	0	0.33		1st stage only
150		113.35		Н	_	4	4	╀	╙	Ц	0	2C/.38	0	0.38		2nd stage only transie
158	<u> </u>	116.95	1	4	. •	┿		╁	<u> </u>	1	1A,1B/.18	20/.67		0.85	•	J data
167	52	165.93	6.0	5	75	0	3000	3	5.2	00	0	0	0	0	No	Ø1A,18 and Ø4,2C
175		172.23				T	Τ				14,18/.24	4,20/.26	0	0.50		
183		180.33				\perp	\perp	\perp	L	Ш	1A,1B/.20	4,20/.41	0	0.61		
191	•	189.33		Ц		\perp	•	L	<u> </u>	L.	1A,1B/.20	4,20/.53	0	0.73	•	•
199	54	156.46	6.0	5	75	, †	3000	1 3	5.2	00	0	0	0	0	No	Constant Ø1A,18, Ø2A,2C
207	,	185.26	\vdash	П		+	Ť	十	T	\vdash	1A,1B/.21	2A,2C/.64	0	0.85		ramped up 3 times
215		200.56		П	\dashv	1	\top	十	†	T		2A,2C/.43	0	0.66		
223		222.16	П	\sqcap	7	1	1	1	†	H	1A,18/.24	2A,2C/.25	0	0.49		
231		235.66		H	\dashv	7	+	1	1	H	1A,18/.24	2A,2C/.52	0	0.76		
239		253.66		П		1	\top	\top	Τ	П	1A,18/.18	2A,2C/.60	0	0.78	1.2	
247	•	280.66			•	1	1	T	•	1	1A,18/.20	2A,2C/.61	0	0.81	No	
255	57	195.11	6.0	, [75	, F	3000	1 2	5.2	o°	0	0	0	0	No	Optimized performance
263	1	207.71		1	ΤÍ	+	T	十	T	İΤ	14,18/.21	2A,2C/.73		0.94	<u> </u>	
27/	 	234.71		H	\dashv	+	+	+	1	H		2A,2C/.60	0	0.92	- -	
279		265.31		H	-	+	+	\top	†	H	1A,18/.21	2A,2C/.36	0	0.57		
287	•	287.81	Ħ	H	-	+	+	1	ŧ		1A,18/.20	2A,2C/.54	0	0.74		
	60	155.69	6	\exists	76	Ŧ	3000	Τ,	5 2	00		0	^	0	No	Variation of fuel schedule
295	80	178.19	١	\dashv	73	+	7	+	7. <u>2</u>	ŀŤ		2A,2C/.64	0	0.85	1	Tarracton of their schedule
303 311	 	186.29	+-	Н		+	+	+	+	H		2A,2C/.65		0.87	 	
311	•	202.49	\vdash	Н	\dashv	+	+	+	†-	${\sf H}$		2A,2C/.65		0.86		
327	 	223.19	\vdash	Н		+	+	+	+	$\vdash \vdash$	1A/.21	2A,2C/.66		0.87	 	
335	 	230.39	H	H	+	+	十	+	t^-	H		2A,2C/.67		0.88		
343		241.19	Н	╁	\dashv	+	+	+	\vdash	H	18/.19	2A,2C/.68		0.87		
351		249.29	H	H	1	+	+	T	†	$\dag \uparrow$	18/.24	2A,2C/.68		0.92	1	
359		258.29	\vdash	H	\dashv	+	十	†	†	十	0	2A,2C/.76		0.76		
367	 - 	264.59	\vdash	╌┼	-+	+	+	+-	+	\vdash	0		0	0.80		

^{*}Reference 10

^{**} Because of insufficient valid engine surface pressure measurements, performance results were not obtained.

a Listings not available.

Table 5. - Continued.

(b) Mach 6 engine performance results.

Page*	Reading				P _T ,	PT	.,	X _{CL}			Inj.1/ø,	10: 2/4	Inj.3/ø ₂		I gni tors	Purpose & Remarks
No.	Number	Time	M	۰	psia	° _R	l	in.	- 1	Œ	Ι	Inj.2/ø2	1117.57.63	97	1, 2, 3	rdipose d Remarks
55	61	178.86	6.	0	750	30	00	36.	7 0	90	0	0	0	0	No	Effect of spike position
63		198.66	\Box		Т		Г		┪	T	1A,18/.13	2A,2C/.36	0	0.49		
72		205.86	П	T		\vdash	T		1	T	1A,18/.15		0	0.64		
81		212.16	\sqcap	\top	╅			П	1	t	1A,18/.15	2A,2C/.61	0	0.76	1.	
90		222.06	1 1	+	+	1	1	1 4	1	t	1A,18/.14	2A,2C/.73	0	0.87	 	
99		231.06	П	\neg	1	 		37.	5 0	0	0	0	0	0	No	Effect of spike position
108		243.66	П	1	1			П	\top	Т	14,18/.30	0	0	0.30	<u> </u>	
117		246.36	П	\top	1		\vdash	\sqcap	+	T	1A,18/.30	2A,2C/.47	0	0.77	 -	
126		251.76	П	1	1		Г	\sqcap	1	T	1A,18/.29		0	0.94		
135		262.56	\Box	1	1				T	T	1A,18/.27	2A,2C/.96	0	1.13	7	Week took coll and
144	•	273.36			+	7	,		٦,	•	1A,18/.26		0	1.41	1 1	High test cell and
	- (1	196 16		\perp	010	1		16	Τ,				Ī			AIM nozz. pressure
153	63	186.15	6.0	4	930	30	00	35.7	Ò		0	0	0	10.00	No	Effect of altitude
161		192.45	╁╌╂	+-	╁	\vdash	\vdash	╀	+	\vdash	1A,18/.24	2A,2C/.56	0	0.80	 	
169		216.75	┥	+-	V 70	\vdash	\vdash	$\vdash \vdash$	+	H	1A,18/.24	2A,2C/.76	0	1.00	}}- -	
177		249.15	┤┤	-	470			$\vdash \bot$	\perp	H	0	0	0	0	 	
103		275.25	_	1	470	Ľ	_		±	_	1A,18/.26	2A,2C/.73	0	0.99		•
193	64	156.11	6.0		750	300	00	35.2	0	0	0	0	0	0	No	Subsonic-supersonic
201		167.81									18/.24	2A,2C/.77	0	1.01		transition
209		202.01		\perp							0	0	3A,38/.85	0.85		
217		239.81			\perp						18/.23	2A,2C/1J1	0	1.34		
225		261.41		\perp					\perp		18/.24	0	3A,3B/.8	1.04		
233	<u> </u>	293.81			•			•	1	L	18/.26	2A,2C/.8	0	1.06	•	
241	65	164.03	6.0	,	750	300	00	35.2	10	▫	0	0	0	0	No	Supersonic combustion
249		174.83	П	+	1			1	ħ	П	1A,18/.23	0	0	0.23		with instrumentation rig,
257		180.23		1	1	П			T	П	1A,18/.24	2A,2C/.34	0	0.58		gas sampling
265		196.43		+	+		Н	\vdash	T	Н	1A,18/.24		0	0.83		
273		201.83	\vdash	+	+	Н	Н		\top	Н	1A.18/.24	2A,2L/.80	0	1.04		
281		218.03		+	+	Н	Н		+	Н	1A,18/.27	2A,2C/.76	0	1.03		
282	+	235.13	1	+	⇟	\vdash	H	+	Ħ	Н	1A,18/.25	2A,2C/.79	0	1.04		
	- (2			Ξ.	150			35.	T	_					N-	
297	69	177.00	6.0	<u>' '</u>	750	300	20	35.2	+	•	0	0	0	0 22	No	Supersonic combustion
305		198.60	├┼	+	+	\vdash	Н	$\vdash \vdash$	+	Н	1A,18/.22	0		0.22	 	with instrumentation rig.
3/3		212.10	-	+	+-	┝╌┤	Н	\vdash	+	Н	1A,18/.23		0	0.48		gas sampling
32/	-	226.50	┝╌┼	+	+	⊦⊣	Ы	$\vdash \vdash$	+	Н	1A,18/.23		0	0.82		
329		256.20	1	+	+-	\vdash	Н	\perp	+-	Н	1A,1B/.22	2A,2C/.69		0.91	- 1 -	
-3/		265.20		土	<u>_</u>		\exists		士	4	1A,18/.23	2A,2C/.79	Ų	1.02		<u> </u>
345	71	160.54	6.0		750	300	00	35.2	3	٦	0	0	0	0	No	Angle of attack perform-
353		171.39							\prod		1A,18/.22		0	0.22		ence
361		174.94	\Box	\perp					\Box	Ц		2A,2C/.31		0.53		
369		193.84			\perp				\perp			2A,2C/.59		0.83		
377		207.34		$oldsymbol{ol}}}}}}}}}}}}}} $	$oldsymbol{L}$				\prod	\Box	1A,18/.24	2A,2C/.81	0	1.05		
385		248.74	\Box	\perp							0	2A,2C/1.33	0	1.33		
393		266.74		\int					\Box		0	2A,2C/.87	0	0.87		
401	<u> </u>	270.34	П						\prod		0	2A,2C/.87	0	0.87		
409		284.74		\mathbf{I}	Ι				\prod		0	2A,2C/.66		0.66		
417	-	285.64	\Box	\top	Π			I		Π	0	2A,2C/.66	0	0.66		I I

^{*}Reference 11

Table 5. - Continued.

(c) Mach 7 component integration and engine performance results.

Dag-4				Γ		P		P		Т		F -					ſ.	
Page*		ding ber	Time	M		P _T	o i a	P _T ,	X _{CL}		inj.1/ø ₁	Inj.2/ø ₂	inj.3/ø ₃	φ _T	Igni 1, 2	tors , 3	Purpose	& Remarks
54	8	8	236.40	7.	25	10	00	3160	36.6	o°	0	0	0	0	2		Exploratory	run
62			245.40					3170			1A,1B/.30	0	0	0.30				
70			261.60					3250		$oxed{\Box}$	1A,1B/.42	0	0	0.42				
78			269.70					3280			1A,18/.55	0	0	0.55				
86			270.60	L	Ц			3270		L	14,18/.57		٥	0.57				
94			271.50	Ļ	Ц			3270		\perp	1A,18/.58		0	0.58				
102			278.70	L.	Ц	Ш	$oxed{oxed}$	3270	—	1	14,4/.16	2A,2C/.70		0.86				
111	_		285.90	L	Ц	Ш	L	3250	\sqcup	ļ.,	14,4/.31	2A,2C/.60		0.91				
120		\sqcup	294.00	<u> </u>	Ц	Ц	L	3200	\vdash	↓_	1A,4/.28	2A,2C/.57	0	0.85				
129			299.40	<u> </u>	\sqcup		_	3150	\vdash	╄	1A,4/.45	2A,2C/.46		0.91	\perp	<u> </u>		
138		4	305.70	-	•		<u>_</u>	3090	-1	╁╌	14,4/.49	2A,2C/.41	ļ.º	0.90	1-1		ļ <u>'</u>	<u> </u>
147	8	39	250.77	7.	4	10	00	1790	36.6	0°	0	0 .	0	0	N	lo	Effect of 1	ow T _{TO}
155			272.37	7.	25			3160	\Box	Γ		2A,2C/.47	<u> </u>	0.79	2			
164			283.17		\Box			3270	\Box	\perp	1A,18/.34			0.89				·
173		$oxed{oxed}$	290.37	L	Ц	Ш		3270	\sqcup	\perp	0	2A,2C/.75		0.75				
181			294.87	L	Ц	Щ		3310	\vdash	╄-	0	2A,2C/.92		0.92				
189			304.77	_	Ц	Ш		3290	\sqcup	\perp	0	2A,2C/.59	1	0.59				
197		\vdash	310.17	╚	Ц	Ш	Ц	3060	\vdash	1_	+	2A,2C/.57		0.89	\square			
206, 232		**	316.47	_			Ц	2720	\sqcup	1		2A,2C/.54	0	0.83	\sqcup			
215,241		**	327.27	7.		Н	Н	2410	$\vdash \vdash$	╀		2A,2C/.54		0.82				
		7	352.47	7.	25	Ľ	<u></u>	3300	-	\vdash	1A,18/.36	2A,2C/.57	0	0.93		<u></u>		
249	9	ю	197.22	7.	25	10	00	3000	36.6	o C	0	0	0	0	N		Optimizatio	n
257		▃▋	206.22	_	Ц				\sqcup	$oldsymbol{\perp}$	1A,18/.48		0	0.48	2			
265			212.52	L	Ц	Ш		\Box	1	╄	1A,1B/.49		0	0.54		L.,		
273			217.02	$oxed{oxed}$	Ц	Ш	L	\Box	1	\perp	1A,1B/.48		0	0.82	\sqcup		ļ	
281			230.52	<u> </u>	Ц	Ш		$\sqcup \sqcup$	\perp	1_	1A,18/.26		0	0.77				
289			235.02	L	Ц				\sqcup	-	1A,18/.79			1.98			Inlet u	nstarted
297			246.72	_	\vdash	Ш	H		1	╀	1A/.51	0	0	0.51	-	ļ		
305		Y	247.62	<u> </u>	<u>_</u>	Ľ	<u>_</u>			上	1A/.55	0	0	0.55		<u> </u>	1	
313	9	19	175.65	7.	25	10	00	3100	36.6	3°			0	0.39	2		Angle of at	tack
321			180.15				L		Ш	L	1A,18/.47	0	0	0.47	2			
329		$oxed{oxed}$	186.45	_	\perp	L_	_		\sqcup	1	0	0	0	0	—	0		·
337		11	190.05	_					\sqcup	1_	14,18/.51		0	0.64	2			
345		\sqcup	203.55	_	L		_		$\vdash \vdash$	╄	1A,18/.52	<u> </u>	0	0.52	\sqcup		·	
353	—	\vdash	216.15	-	\vdash	\vdash	-		$\vdash +$	+-	18/.27	4,20/.34	0	0.61	\vdash		L	· ·
361			224.25	 	<u> </u>		H			╀	18/.28	4,20/.50	0.	0.78	╂─┤			
369		\vdash	226.95	H	\vdash	\vdash	H		├┼	+	18/.28	4,20/.45	0	0.73	┝╌┤			
377 385	-		229.65	 	Ŀ				$\vdash \bot$	╀	18/.33	4,2C/.39 2C/.41	0	0.72				
			235.95	_	Ψ		_			上								
393	- 9	92	186.87	_		10	00		36.6	100	+	0	0	0	N.		Supersonic of	combustion mentation rig,
401			205.77	7.	29			2850		\perp	1A,18/.48	1	0	0.72	2	·	gas samplin	g and O2
409			227.37					\Box	\Box	igspace	1A,1B/.50		0	0.93	\Box		content eff	ect
417		1_	248.07	_	•	\sqcup	L		$\vdash \downarrow$	╀	18/.33	4,20/.58	0	0.91			ļ	
425		\downarrow	290.37		25		_	3000	$\vdash \downarrow$		1A, 15/-47		0	1.12	┝╌┤		<u> </u>	
433	<u></u>	Y	312.87	7.	25	L	_	3000		L	1A,1B/.36	4,20/.49	0	0.85				<u></u>

^{*}Herein

^{**} Recomputations were made with surface pressure substitutions

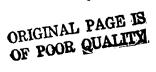


Table 5. - Continued.

(d) Mach 5 component integration and engine performance results.

Page*					1	۴,	PTO	Y	•								
No.	Rea Nun	idi ng iber	Time	M _o		o osia		^c	L'	œ	1nj.1/ø ₁	Inj.2/ø ₂	Inj.3/ø ₃	Фт	Igni i	tors	Purpose & Remarks
54	9	93	134.03	5.1	1	+20	2100	39	. 2	00	0	0	0	0	No		No fuel injection
62		l	142.13	П	T			T	Г		0	2A/.29	0	0.29	2		2nd stage only
70			150.23	П	Т	T	П	Т	П		0	2A/.31	3A,38/.25	0.56	ГΤ		Subsonic combustion
78			158.33	П	T	Т		T	T		0	0	3A,38/.60	0.60	\Box		and 0, content effect
86			162.83					T	Γ		0	0	3A,3B/.71	0.71			
94			174.53					T	Г		0	0	3A,3B/.49	0.49			
102			182.63	•	\perp	<u>. </u>	I		▼		0	0	3A,3B/.35	0.35			•
110	9	4	134.14	5.1	14	20	2230	35	. 2	00	0	0	0	0	No		Subsonic combustion
118		i -	140.44	T	\top	T	Ť	+	T		0	2A/.49	0	0.49	2		
126			150.34		†	T	\Box	+	T		0	2A/.49	3A,38/.47	0.96			
134			157.54	\sqcap	+-	1	- -	+	T		0	0	3A,38/1.03		 		
142			163.84	\sqcap	+	T		+	T		0	0	3A,3B/1.19		 - 		
50			180.04		\top			1	T		0	0	3A,38/.59	0.59			
158			214.24	\sqcap	3	00	2940			П	0	2A/.53	0	0.53		$\overline{\ }$	Effect of T _{TO}
166			215.14	$\vdash \uparrow$	1	Т		+-		Н	0	2A/.53	0	0.53	+	士	
174			218.74	\sqcap	1-		\vdash	+-	T	Н	0	2AV.54	3A,38/.5	1.04	-	T	High test cell and
83			231.34	1	†			†-	T	Н	1A.1B/-15		0	0.15			AIM nozz. pressures
91			233.14		Τ			T	T		1A,18/.25	0	0	0.25			
99			234.04			•	•	1_	•		1A,1B/.27	.0	0	0.27		•	
30	. 9	-	129.55	5.2	Τ,	00	2430	35	,	0	0	0	0	0	No		Supersonic combustion
207			140.35	5.1	+-	1	3080	4		\dashv	1A,1B/.16		0	0.16	2		Supersonic Compustron
215			160.15	7.1	+	+	2940	+	Н			2A,2C/.68	0	0.86	i		
231			169.15		╁╌		2540	+-	H			2A,2C/.83	0	1.02	 		
			189.85	+	╁	+	-	+-	╁┤	-	0	2A,2C/.99		0.99	1		
239			196.15	1	+-	+	+	╁╴	Н		0	2A,2C/.86	0	0.86	1		
47			204.25	+	+-	+	-	+-	Н		0	2A,2C/.71	0	0.71	 		
55 63			211.45	+	+-	T	-	+-	Н		-0	2A,2C/.58		0.58	-		
	_		217.75	+	+-	+1	-	+-	Н	-	0	2A,2C/.70		0.70		-	
271			228.55	+	╁╌	+	+	+-	Н	-+		2A,2C/.63	0	0.85	\vdash	_	
87	-		241.15	\dashv	╁╌		+	╁╴	H	-	0	0	0	0	No		
95	\neg		252.85	+	1	20	2800	╁╴	Н	\dashv		2A,2C/.70		0.88	2		
03			289.75	+	+-	10	2890		Н	\dashv	0	2A,2¢/.86	0	0.86	ΙŤ	_	AIM nozz. press. hi
111			310.45	+	+	20	2230	+	H	-+	0	2A,2C/.66		0.66	-	\dashv	Effect of T _{TO}
719		,	3:7.65	+	┷-	20	2230		H	-	0	2A,2C/.51	0	0.51	-+	\dashv	
			134.44	5.1	-	20	2230		긁	,	0	0	0	0	No		Angle of attack perform-
27	- 9	-	141.64		+-	+	1230	1"	H	'	0	2A/.38	0	0.38	2		ance
36	\dashv		150.64	+	+-	\vdash	+	+-	Н		0	2A/.45	3A,38/.38	0.83	-	\dashv	aug .
44			165.94	+	+-	╂╼┨		┼~	H		0	0	3A,38/.87		 	÷H	
352	_		172.24	+	╁╌	H	+	+-	Н	+	0	0 .	3A,3B/.59		-+	-	
60			180.34	+	+-	H	\pm	+-	H	\dashv	•	0	3A,38/.43		\vdash	\dashv	
68			244.24	+	1,	00	2925	+-	Н	-+	0	0	0	0.45	No	-	
76			264.04	+	-	20	2230		Н	+	1A,18/.10		0	0.10	2		Fuel flow meas.
84			274.84	+	+*		2230	+-	⊢∤		1A,18/.10		0	0.70	- 2	+	malfunction; lA
92		\longrightarrow		+		H	+	+-	${\mathbb H}$		1A,18/.20		0	0.20	2	-{	flow only indicated
00			275.74	_	 -	╁┤	-	╀-	$\vdash \vdash$	-1		0	0	0.20	No	<u> </u>	TION ONLY INGICATED
08		.—	294.64	+	┼	╀	+	├	⊦∔	-+	0	0			2		
47	1		313.54	_ 1	1 _	<u> </u>		1	$oldsymbol{L}oldsymbol{L}$	_1	0		3A,38/.77	0.77		1	High test cell and

*Reference 12

Table 5. - Concluded.

(d) Concluded.

rΛ 1	, ,					_		_	_	_	nd	re
Remarks	stion with	n rig and	robes								cell a	pressu
lgnitors 1, 2, 3 Purpose & Remarks	Subsonic combustion with	instrumentation rig and	gas sampling probes								High test cell and	AIM nozz. pressures
ors 1												•
Igni tors 1, 2, 3	S.	2										•
4	0	0.90	0.56	0.50	6.67	98.0	0.93	0.77	0.74	0.90	1.07	1.08
Inj.1/ø ₁ Inj.2/ø ₂ Inj.3/ø ₃	. 0	34,38/.49 0.90	34,38/.24 0.56	34,38/.50 0.50	34,38/.67 0.67	34,38/.86 0.86	34,38/.43 0.93	34,38/.34 0.77	34,38/.74 0.74	34,38/.90 0.90	3A,38/1.07 1.07	3A,38/1.08 1.08
Inj.2/ø ₂	0	2A/.51	2W.32	0	0	0	2A/.50	24.43	0	0	0	0
Inj.1/ø ₁	0	0	0	0	0	0	0	0	0	0	0	0
ð	္0											-
^χ cι. in.	35.2											•
PTo' XCL'	2100 35.2	2200										•
P. To	210					•	420					-
z ° .	5.1											•
Time	135.71	156.41	r60.91	182.51	14.102	224.81	1252.71	19:1/2	16.295	317.51	322.01	325.61
Reading	97	-										-
* Page No.	425	433	442	451	459	467	476	485	484	502	210	518

*Reference 12

```
0000000 PROCDEF CO33
0000100 KDOSEL 60, 65, 67, 83, 84, 85, 86, 87, 88, 91, 92,123,124,148,154,156,158,160,162,164
0000200 KDOSEL 165,166,168,171,172,174,175,176,180,181,182,183,185,191,206
0000300 KDOSEL 208,212,226,228,230,231,236,239,240,241,244,248,249,290,292
0000400 KDOSEL 305,306,507,308,309,310,311,312,313,318,315,317,318,319
0000500 KDOSEL 310,321,322,323,324,325,326,327,328,329,330,331,332,333,334
0000500 KDOSEL 355,336,337,338
0000700 KDOSEL 399
0000800 QUALIFY AINLETT
0000300 AT 3(2);SET VAL(11, INITRO)=.73448,VAL(11, IOXY)=.28552;DISPLAY VAL(11, INITRO),VAL(11, IOXY)
0001000 QUALIFY STARRS
    CO33
CO33
CO33
   CO33
   CO 33
    CO33
   CO33
   CO 33
CO 34
                                                0001100 AT 320(2);DISPLAY 'INPUT PSI(1,1), THEN TYPE GO'
                                             0000000 PROCDEF CO34

0000100 KDOSEL 60, 65, 67, 84, 85, 86, 87, 88, 92,123,124,188,134,156,158,160,162,164

0000100 KDOSEL 166,168,171,172,174,176,180,181,182,183,186,191,195,199,201

0000100 KDOSEL 206,208,212,226,228,230,231,236,240,241,244,248,249,252,290,292

0000400 KDOSEL 305,306,507,308,309,310,311,312,313,314,315,316,317,318,319

0000500 KDOSEL 320,321,322,323,324,325,326,327,328,329,330,331,332,334,335

0000500 KDOSEL 336,337,338

0000700 KDOSEL 336,337,338

0000700 KDOSEL 339

0000800 QUALIFY AINLETT

0000900 AT 3(2);SET VAL(11,1NITRO)-.73448,VAL(11,10XY)-.26552;DISPLAY VAL(21,1NITRO),VAL(11,10XY)
    CO34
   CO34
   CO34
   CO34
CO34
CO34
   CO 36
                                                0000100 KDOSEL 60, 65, 66, 67,123,124,144,154,156,158,160,162,164,166,166,171,172,174,181 0000200 KDOSEL 182,186,191,195,198,206,208,218,228,230,231,236,260,241,244 C000300 KDOSEL 248,249,252,280,280,292,284,305,310,312,313,314,315,326 C000400 KDOSEL 395
   CO36
CO36
                                             CO36
     CO 36
   C038
    CO38
   C03&
   CO38
   CO 3.6
   CO 52
   CO 52
                                              0C00400 QUALIFY AINLETT
0C00500 AT 3(2);SET VAL(11, INITRO)=.73448, VAL(11, IOXY)=.26552;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
000C000 PROCDEF COS4
   CO 52
   COSA
                                              000C000 PROCDEF C05%
C000100 KDOSEL 65, 65, 67,12%,137,139,1%1,15%,165,168,178,181,182,195,199,200,201,206,226,230
C000202 KDOSEL 2%9,252,268,289,290,292,29%,305,313,31%,315,319,320,329,399
C000000 QUALIFY AIMLETT
DC005CC AT 3(2);SET VAL(11, INITRO)=.73%%,VAL(11, IOXY)=.26552;DISPLAY VAL(11, INITRO),VAL(11, IOXY)
C000000 PROCDEF C057
   CO54
CO54
   COSA
  CO54
                                            COOOOOO PROCDEF CO57

OCOLOR DOSEL 52, 65, 66, 7%,12%,137,139,158,160,168,172,179,181,182,183,187,190,195,199

COCOZOO KDOSEL 529

OCOCOZOO KDOSEL 529

OCOCOZOO KDOSEL 599

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   CO57
   COS 7
   CO 5 7
  C060
   6003
  C060
  C060
                                         0000500 QUALIFY AINLETT
0000600 AT 3(2);SET YAL(11, INITRO)=.73613, VAL(11, IOXY)=.26387;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
0000000 PROCCEF CO51
0000100 KDOSEL 62, 55, 68, 74,124,137,139,158,160,168,172,179,181,182,183,187,190,195,199
0000200 KDOSEL 201,206,226,230,248,249,252,289,290,292,294,305,313,314,315,319,320
0000300 KDOSEL 329
0CD000 KDOSEL 329
0CD000 QUALIFY AINLETT
0000600 AT 3(2);SET VAL(11, INITRO)=.73928, VAL(11, IOXY)=.26072;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
0C00000 PROCCEF CO53
000100 KDOSEL 199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315,319
0C00100 KDOSEL 199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315,319
0C00100 KDOSEL 399
0C00500 QUALIFY AINLETT
0C00600 AT 3(2);SET VAL(11, INITRO)=.7728, VAL(11, IOXY)=.2276;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
0C00000 PROCCEF CO64
0C00050 KDOSEL 62, 65, 66, 74
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 197,199,201,206,226,230,248,249,252,289,290,292,294,305,313,314,315
0C00100 KDOSEL 62, 65, 66, 74,137,139,181,182,183,187,180,195,197,199,201,206,226,230
0C00100 CDOSEL 248,252,289,290,292,294,305,313,314,315,320,321,329,399
0C00100 CDOSEL 248,252,289,290,292,294,305,313,314,315,320,321,329,399
0C00100 CDOSEL 248,252,289,290,292,294,305,313,314,315,320,321,329,399
0C00100 CDOSEL 62, 65, 66, 74,137,139,181,182,183,187,190,195,197,199,201,206,226,230,248,252
  C060
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C059
                                              0000000 PROCDEF CO69
                                              0000100 KDOSEL 62, 65, 66, 74,137,139,181,182,183,187,190,195,197,199,201,206,226,230,248,252
0000200 KDOSEL 289,290,292,294,305,313,314,315,320,321,322,329,399
0000300 AT 5(2),3ET TATC(11,1N1TRO)-.76479,VAL(11,10XY)-.23521;DISPLAY VAL(11,1N1TRO),VAL(11,10XY)
C069
```

```
0000000 PROCDEF C071
0000100 KDOSEL 53, 62, 65, 66, 74,124,137,139,158,260,172,179,181,182,183,187,190,195,197,199
0000200 KDOSEL 201,206,226,230,248,249,252,289,290,292,294,305,313,314,315,320,321,322,329,399
0000500 QUALIFY AINLETT
0000600 AT 3(2);SET VAL(11,1N1TRO)=.75452,VAL(11,10XY)=.24548;DISPLAY VAL(11,1N1TRO),VAL(11,10XY)
0000000 PROCDEF CO88
    CO71
CO71
CO71
     C071
    C071
                                                      0000000 PROCDEF CD88
0000100 KDOSEL 19, 22, 23, 54, 55, 60, 62, 64, 67, 74, 95,124,137,139,157,158,168
0000100 KDOSEL 19, 22, 23, 54, 55, 60, 62, 64, 67, 74, 95,124,137,139,157,158,168
0000200 KDOSEL 182,183,187,190,195,197,199,206,226,227,230,235,241,248,249
0000400 KDOSEL 250,252,278,289,280,282,284,305,313,314,315,320,321,329,349
0000500 KDOSEL 353,366,367,368,369,370,374,375,378,379,382,388,394,395,399
0000800 QUALIFY AINLETT
0000900 AT 3(2);SET VAL(11, INITRO)=,75328,VAL(11, IOXY)=,24672;DISPLAY VAL(11, INITRO),VAL(11, IOXY)
0001100 AT 360(3);SET DRAGEX==0.5=Q0AC;DISPLAY DRAGEX,DRAGEX=PSIATM, DRAGEX = -0.5=Q0+AC*
0001200 QUALIFY CONVTA
0001300 AT 0;SET MY(65)=MY(53),MY(66)=MY(53);DISPLAY MY(53),MY(65),MY(66)
0001400 AT 0;SET MY(65)=MY(53),MY(66)=MY(53);DISPLAY MY(53),MY(65),MY(66)
0001400 SETPS 123,0,690
0000000 PROCDEF CD89
     CO8 8
    CO88
     COSE
    C088
  C088
    COSS
  C088
C088
                                                       0000100 PROLEF COSP

0000100 PROSEL 54, 55, 60, 62, 64, 67, 74, 95,124,137,139,157,158,160,165,166,169

0000200 KDOSEL 172,175,176,179,181,182,183,187,190,195,197,199

0000300 KDOSEL 210,225,224,226,227,230,235,248,249,250,252,289,290,292,294

0000400 RDOSEL 305,313,320,321,329,399

0000600 QUALIFY AINLETT
    CD89
     C089
    C089
     C089
    C089
                                                       0000700 AT 3(2);SET VAL(11, INITRO)=.75188, VAL(11, IOXY)=.26852;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
0000800 QUALIFY CONVTA
0000900 AT 0;SET MY(65)=MY(53), MY(66)=MY(53);DISPLAY MY(53), MY(65), MY(66)
0001000 SETPS 123, 0.690
  CO89
  CO89
CO89
                                                  000900 AT 0.SET W(65)=WY(53), WY(66)=WY(53); DISPLAY MY(53), MY(66)
001000 SETPS 123,0,690
0000000 PROCCEE COS9
0000100 KDOSEL 197,181,182,183,187,190,195,197,199,202,203,206,207
0000200 KDOSEL 197,181,182,183,187,190,195,197,199,202,203,206,207
0000303 KDOSEL 208,210,215,724,226,227,230,235,248,249,250,252,273,289,290
0000500 KDOSEL 199
0000500 QUALIFY AINLETT
0000700 AT 3(2);SET VAL(11,1N1TRO)=,7389,VAL(11,10XY)=,2611;DISPLAY VAL(11,1N1TRO),VAL(11,10XY)
0000800 QUALIFY CONVTA
0000000 PROCCEF COS1
0000100 KDOSEL 54, 55, 60, 62, 64, 67, 74, 96,124,137,139,148,157,158,160,165,172
0000200 KDOSEL 54, 55, 66, 62, 64, 67, 74, 96,124,137,139,148,157,158,160,165,172
0000000 ROSCEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 KDOSEL 175,176,179,181,182,183,187,190,195,197,199,206,238
0000000 GUALIFY AINLETT
0000700 QUALIFY CONVTA
0000000 QUALIFY CONVTA
0001000 GUALIFY CONVTA
0001000 GUALIFY CONVTA
0001000 GUALIFY CONVTA
0001000 GUALIFY CONVTA
0001000 GUALIFY AND?
0000000 ROCCEF COS2
0000000 ROCCEF COS2
0000000 ROCCEF COS2
0000000 AT 360(3);SET DRAGEX=-0.5=QOAC;DISPLAY MY(51),MY(55),MY(55),MY(56)
0000000 QUALIFY AND?
0000000 AT 360(3);SET DRAGEX=-0.5=QOAC;DISPLAY MY(53),MY(55),MY(56)
0000000 AT 360(3);SET DRAGEX=-0.5=QOAC;DISPLAY MY(53),MY(55),MY(56)
0000000 ROCCEF COS2
0000000 AT 360(3);SET DRAGEX=-0.5=QOAC;DISPLAY MY(53),MY(55),MY(56)
0000000 AT 360(3);SET VAL(11,INITRO)=,65570b,VAL(11,IOXY)=,384296;DISPLAY VAL(21,INITRO),VAL(21,IOX)=,0000000 AT 360(3);SET VAL(11,INITRO)=,65570b,VAL(11,IOXY)=,384296;DISPLAY VAL(21,INITRO),VAL(21,IOX)=,0000000 AT 360(3);SET VAL(11,INITRO)=,65570b,VAL(11,IOXY)=,384296;DISPLAY V
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                                                       0000500 QUALIFY AINLETT
0000500 AT 3(2):SET VAL(11, INITRO)=,85570%, VAL(11, IOXY)=,3%%298;DISPLAY VAL(11, INITRO), VAL(11, IOXY)
0000700 TUNNOPT 3
00007000 PROCDEF CO9%
  CO93
  C093
C093
C094
                                                      0000000 COMACHS .

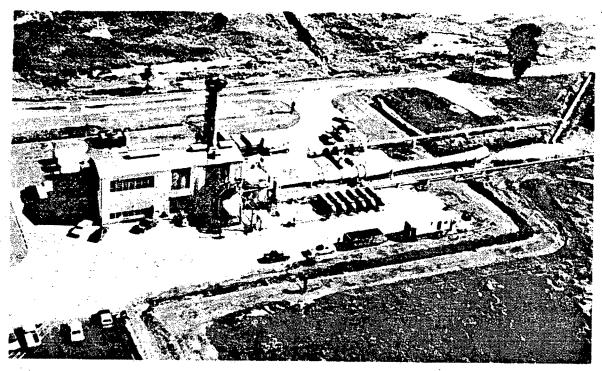
0000600 QUALIFY AINLETT

0000600 AT 3(2);SET VAL(11, INITRO)=.76284, VAL(11, IOXY)=.23716;DISPLAY VAL(11, INITRO), VAL(11, IOXY)

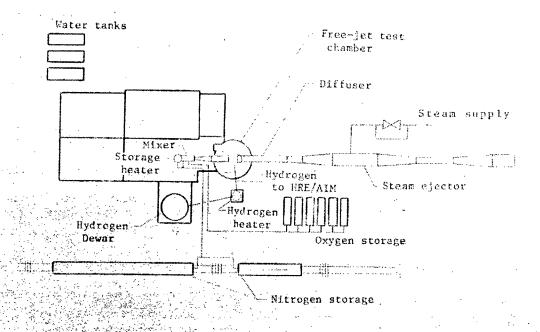
0000800 TUNNOPT 3

0000000 PROCDEF COSS

0000100 COMACHS
     C094
    C094
  C094
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COSS DODOIGO DIALIFY AINLETT
COSS DODOIGO AT 360(3);SET VAL(11, INITRO)=.77086, VAL(11, 10XY)=.2291b;DISPLAY VAL(11, INITRO), VAL(11, 10XY)
COSS DODOIGO DIALIFY AINLETT
COSS DODOIGO AT 360(3);SET VAL(11, INITRO)=.77086, VAL(11, 10XY)=.2291b;DISPLAY VAL(11, INITRO), VAL(11, 10XY)
COSS DODOIGO DIALIFY AINLETT
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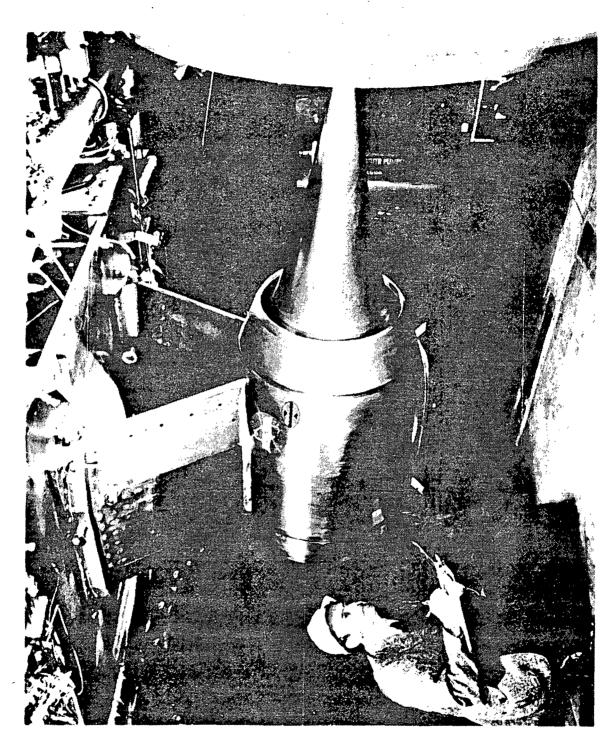


(a) Hypersonic Tunnel Facility (HTF).

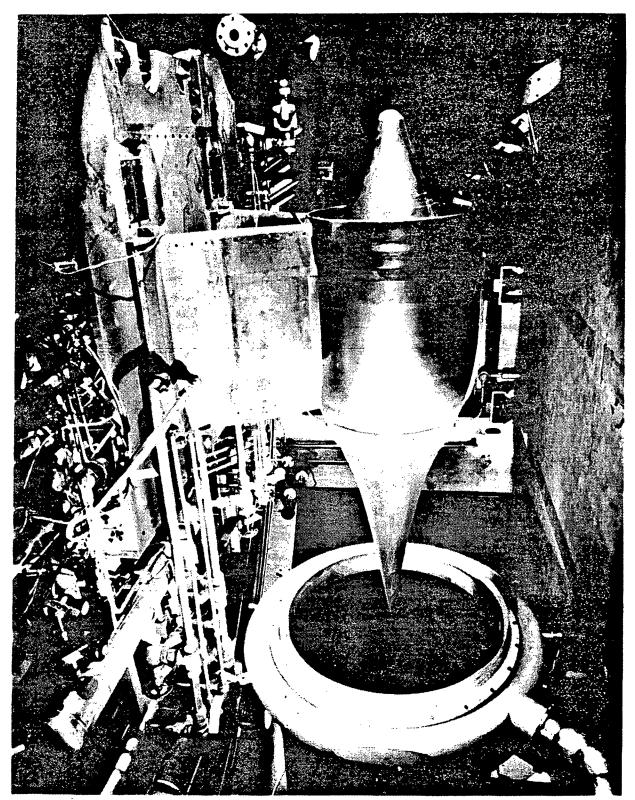


Schematic layout of the NASA - Lewis - Plum Brook Hypersonic Tunnel Facility(HTF).

Figure 1. - NASA - Lewis Research Center's Plum Brook Station Typersonic Tunnel Facility (HTF) and the Hypersonic Research Engine/ Aerothermodynamic Integration Model (HRE/AIN) Installation.

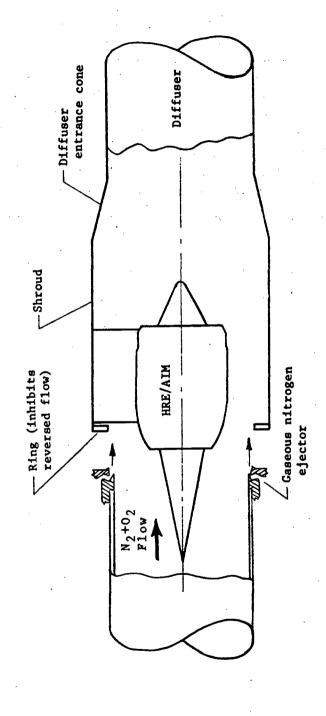


(c) HRE/AIM partically installed: protest.



(d) HRE/AIM partically installed; Mach 5, 5, and 7 post test.

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(e) Schematic of HRE/AIM test section located in the free-jet test chamber of the HTF.

Figure 1. - Concluded.

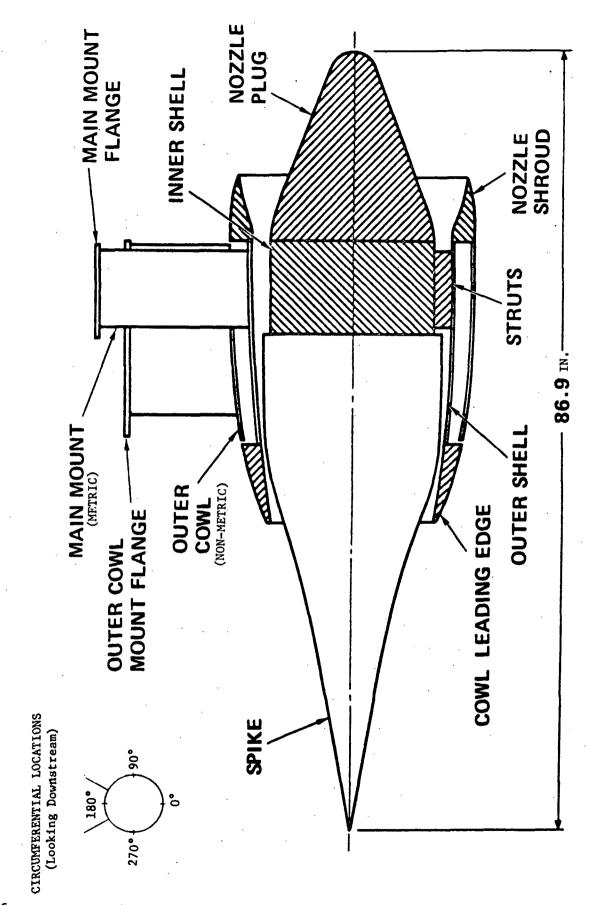
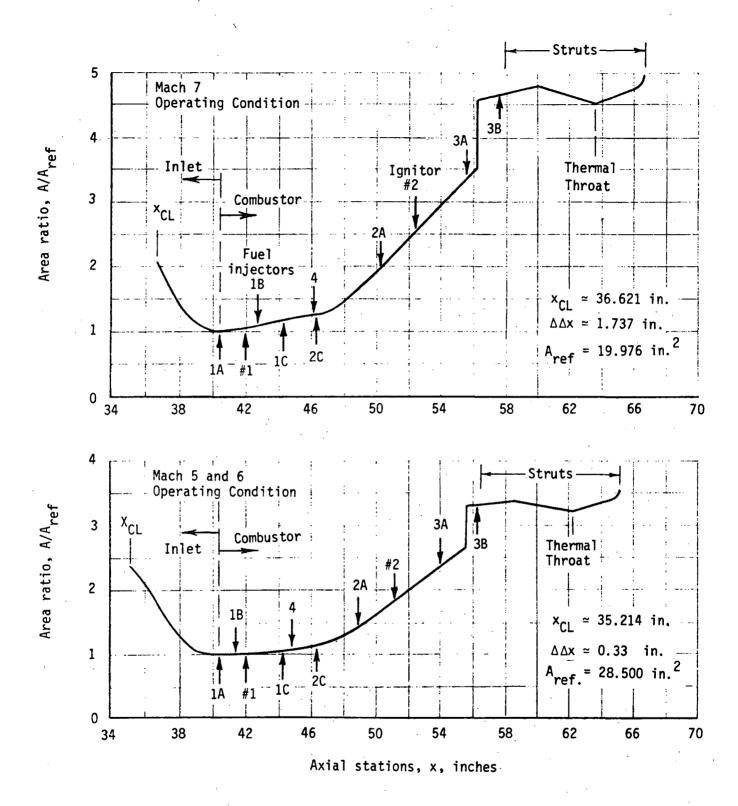


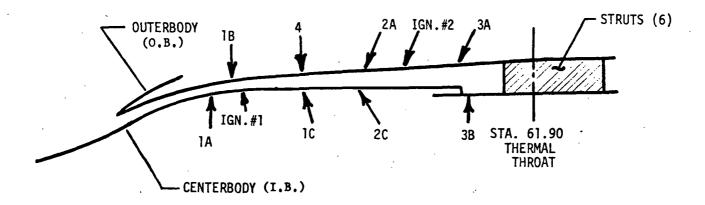
Figure 2. - General Configuration of the AIM



(a) Combustor area ratio distributions

Figure 3. - HRE/AIM combustor information.

COMBUSTOR CONFIGURATION



(Mach 6 position, $x_{CL} = 34.884$ in.)

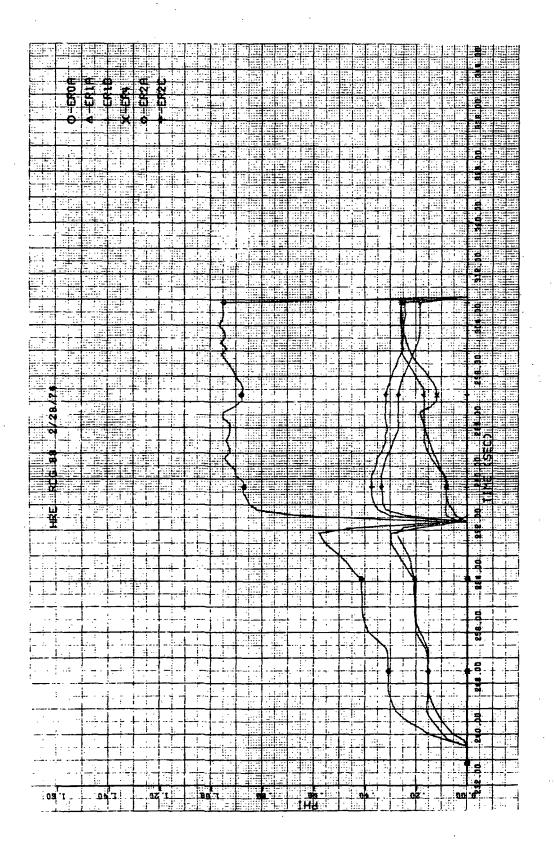
Injector	Number of Injectors	Diameter, in.	Injection	S/4	v in	Location
2110 000	111300013	Dianeter, in.	raigic ; acg.	<u> </u>	<u> </u>	Locucion
1A	37	0.119	-90	13.1	40.5	I.B.
18	37	0.119	90	13.9	41.25	0.B.
10	37	0.119	106	13.5	44.5	I.B.
4	37	0.119	90	14.2	44.5	0.B.
2A	60	0.095	67	11.4	48.5	0.B.
20	. 60	0.095	119	10.6	46.5	I.B.
3A	114	0.090	65	7.0	53.75	0.B.
3B	102	0.095	90	6.3	55.9	I.B.

IGNITOR	PARAMETERS

Ignitor	<u>x, in.</u>	· <u>{</u>	Circum	ferent	ial lo	cation	<u>s</u>	Injection Angle ^a , deg.	Location
ıc	42.00	55	110	165	230	290	350	94.5	I.B.
2	50.98	40	100	 -	220	240	280	60.0 ^b	0.B.

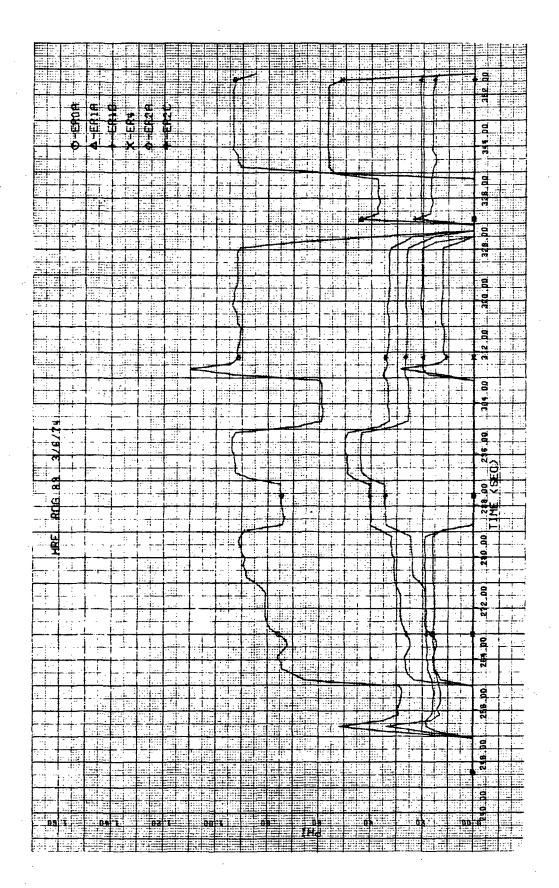
- a. With respect to AIM centerline.
- b. Also looking upstream, ignitors #2 are inclined 30° clockwise.
- c. Plug welded prior to reading 57.
 - (b) Combustor configuration and parameters.

Figure 3. - Concluded.



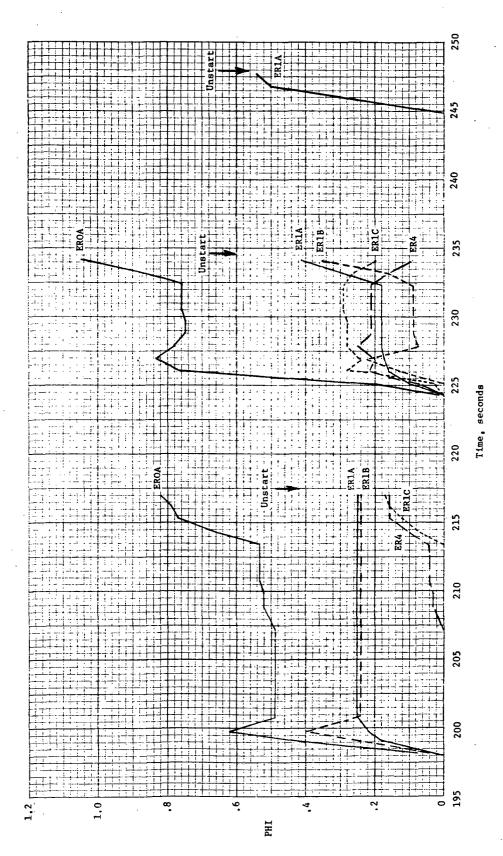
(a) Reading 88 - Measured Equivalence Ratio, ϕ

Figure 4. - HRE/AIM fuel equivalence ratio; Mach 7 component integration and performance results.



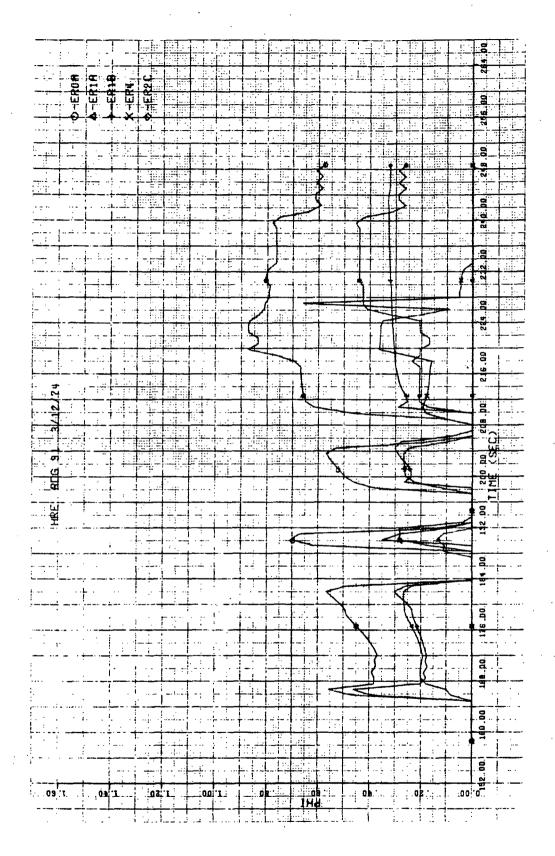
(b) Reading 89 - Measured Equivalence Ratio, 9

Figure 4. - Continued.



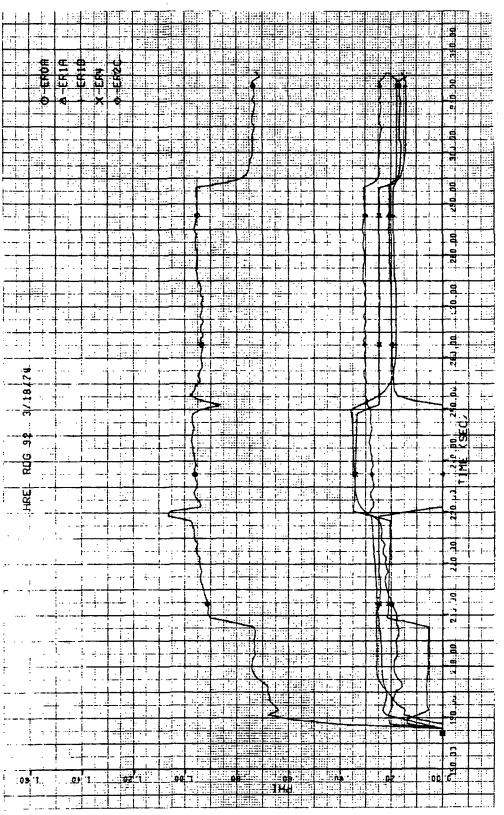
(c) Reading 90 - Measured equivalence ratio, ϕ

Figure 4. - Continued.



(d) Reading 91 - Measured Equivalence Ratio, Ø

Figure 4. - Continued.



(e) Reading 92 - Measured Equivalence Ratio,

Figure 4. - Concluded,

Reading 88

t = 236.40 sec.

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7.2	- - - -	7208	2645 858	2645		2645	2618	2618	2618	2618	2616	2612	2610	2602	2599	2599	2587	2584	2579	2573
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HOWANDIBOOK BALOWS

DAMJRT PFHFORMANCE

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Reading 88

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Combustor pressure distributions indicate the injected fuel did not ignite.

THE BEDGIOOF RACE 7.2 DIE GORANGE TIE SYKOSA AND SAKONA HTUCK # 120 READING # 008c

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READING B 0.348 PLOCK B 178 TIME B 274,701 AACH 7.2 PT B 999,249 TT B 3266.3 CORAG

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	COPBUSTON EXITATION OF THE PROPERTY OF THE PRO	66.479	(31)	. 3	46.220	e.	

ENGINE DEPENHANCE

Reading 88

t = 285.90 sec.

Combustor pressure distributions indicate the injected fuel did not ignite,

Fuel ignition during the second fuel schedule appeared to occur at about time 290 seconds when the fuel flow rate from injector 4 reached a peak prior to a fuel flow decrease (see fig. 6(a)).

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v o	1.623	4,131	1.623	2,131	1.890	1.890	1.970	2.061	2.064	2.050	3.060	660*2	. 060	2.155	2.151	2.158	2.171	2,242
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IVAC	136.2	148.0	148.0	150.7	154.2	150.0	166.1	166.1	166.3	173.6	178.1	179.2	180.B	181.3	196.2	189.5	189.6	190.0	196.2
e	9.374	8.978	846	7.540	5.477	044.98	4,118	6.1.19	4.210	1.039	18.600	17.618	7.061	9.9	3.930	0.867	1.083	0,925	77990
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ø	2,292	2.399	2.399	2.412	2.429	2.447	2,550	2.550	2.532	2.563	2.568	2.588	2,583	2.580	2,596	2.694	₹.690	2,695	2.694
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t = 272.37 sec.

Injected fuel did not ignite.

The first occurrance of fuel ignition occurred at time 309 seconds when fuel flow from injectors 1A and 1B was reinitiated and 1B flow "spiked" (see fig. 6 (b)).

TVAC PIT ETAC 2297 40.464 154.4 0.32 0.02 2354 50.916 158.2 0.32 2483 56.130 167.7. 0.18 2444 55.940 145.0 U.18 2420 54.477 163.4 0.1A 2564 51.981 159.6 0.18 2357 40.584 156.1 0.32 2472 55.242 166.9 5.627 196.7 0.882 211.3 6.121 196.4 4.491 JA6.0 2044 50.225 168.0 2503 51.797 170.0 2903 12.324 170.0 2354 SC.480 198.3 2313 43.072 155.5 01,002 154.he 41.404 2846 2115 3152 3152 231.4 847) 1.2978 28.908 2652 74) 1.3967 28.908 861 7.247 6242 1.875 0.6581 14.727 6.9912 912 2.132 0.05801 106727 re9412 28.909 2610 26.908 1764 2.815 4964 1.894 0.72878 14.727 0.0789 1.2911 28.908 1732 2.905 5031 1.894 0.66253 14.727 A.A668 7.109 6231 1.825 0.06320 16.045 c.9912 28.908 2610 28.408 2572 0.465 1197 1.972 0.66253 14.727 0.0888 26.974 2680 26.974 1876 2.627 4929 2.022 0.73279 14.A10 n.6789 26.914 2658 26.914 1874 2.577 4833 2.012 0.73549 14.81° n.r7f6 26,902 2650 P6,901 1F16 2,716 4931 2,019 0,72994 14,810 6,6792 26,990 2646 26,899 1846 2,627 4851 2,024 0,722A7 14,810 0,4800 25.544 2691 25.544 1906 2.542 4844 2.157 0.67719 14.A77 0.A848 2.593 4843 2.147 0.67658 14,877 0.0858 928) 1.3041 25.515 2666 46A) 1.3448 25.515 2086 2.040 4256 2.140 0.61177 14.477 0.0949 20.899 1805 2.718 4905 2.057 0.68190 14.810 0.6848 831) 1.3042 25.492 2471 367) 1.3628 25.492 1874 2.479 4832 2.145 0.67366 10.817 0.6842 25.491 2661 25.491 2011 2.212 4489 2.135 0.62301 14.877 0.4932 25.491 2659 25.490 2050 2.11% 43%2 2.1%0 G.A.1562 14.877 0.0943 37/8 8 / 8 اب دو د 28,908 2651 26,908 2650 0,347 MURI ANDS LATER BUNK DAKE 723.7(A47) 1.2878 28.909 2652-52.31 77) 1.3971 28.908 A77 1,3039 25,499 2674 1,3626 25,498 1875 1.2911 1.2944 1,2964 0 6 725.71 Au7) 1.2472 707.11 A31) 1.2899 1.2940 1.3017 1.3562 1,2994 1,3512 578.16 822) 1.3051 303.26 449) 1.3480 1,2967 723.7(847) -55.0(74) 82A) 30S) 815) 853) 855) 37() 8253 35^A3 51A) 833) 690.6r 815) 198.0r 327) **3**81) 679.66 823) 284.16 430) 190.0f 815) 847) A22) 690.61 686.87 ZZU.37 697.66 94.00 112.16 678.16 11,462.3179 0,600 10,488.3109 -LET THROAT TUNNEL 18175 11,462 3175 994.709 3175 15 2805 0 9.015 128n 100.966 3067 87.960.2973 10,616 3121 315.662 3067 10.449 1331 315,862.306 108.133 401.718 10.607 99.693 5,224 5.314 118.925 249.551 219.176 6.571 4.871 5.211 107,092 10.715 12.446 124.907 12.271 MIND TUNKEL OMBUSTOR 00000

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t = 290.37 sec.

Injected fuel did not ignite,

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READING # 0089 BLOCK # 114 TIVE # 200,371 FACH 7,2 PT # 994,749 TT # 3271,0

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RAPURT PERFUGGRAPP

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t = 294.87 sec.

Injected fuel did not ignite.

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PLOCK # 119 TIVE # 244.875 NACH 7.2 PT # 994.449 TI B 5308.5 READING # 0089

ETAC T d 201.3 216.7 2450 59,191 172,4 2470 50.884 173.8 2070 18,285 175.6 2440 55.048 171.7 2308 52.458 168.5 2238 3t.425 157.5 6.07¢ 201.1 0.974 201.1 2450 55.174 172.4 2346 47.843 165.1 5.571 108.0 2416 53.762 40.030 41.830 2338 46,462 G 3080 3119 2861 3119 ROW I R 1,2839 28,909 2703 1,3972 28,908 A84 7,231 A390 1,838 0,35610 14,211 A,9890 887) 1.2839 28.909 2763 81) 1.3976 28.908 90^ 7.090 8379 1.83% 0.08124 15.512 0.9890 28,909 2665 28,408 1904 2,537 4868 1,928 0,69342 14,211 0,0800 925 2.145 0.05610 14.21.1 0.9890 28.907 2576 0.382 1023 2.145 0.05124 19.512 n.9890 28,408 1840 2,744 5050 1,915 0,70321 14,211 0,0789 26,909 2674 26,906 1807 2,634 5122 1,915 0,63929 14,211 0,0868 28,908 2673 26,908 2635 0.869 1237 1.989 0.63929 14,211 n.n668 20,909 2673 20,900 1641 2,743 5049 1,915 0,70312 14,211 n,0789 26,909 2671 26,908 1850 2,713 5019 1,916 0,70572 14,211 0,6786 28.900 2657 28.900 1956 2.400 4705 1.941 0.65430 14.211 0.0848 2015 2.217 4468 1.957 0.588n6 14.211 n.n941 2020 2040 2,133 4351 1,943 0,36829 14,211 0,0976 2667 1878 2,630 4939 1,922 0,70039 14,211 0,0792 28.909 1962 2.384 4879 1.943 0.64628 14.211 0.0858 2054 1964 2.379 4672 1.943 0.64349 14.211 0.0862 2643 2001 2.261 4523 1.954 0.59312 14.211 n.n932 28,908 2634 28,908 2033 2,162 4394 1,960 0,58437 14,211 0,0949 AIAC 7 . **∨** 6. 1 **7** C I 2701 GAMMA MOLYT GONY 28.909 26.909 26.908 26.908 28.908 26.908 1.2827 28.407 1,2827 1,2861 1.2860 1,2863 1,2861 1.2861 1.2866 1,2868 1,2874 1,2876 1,2869 1,3351 1,2892 450) 1,2896 456) 1,3332 1.3401 1,3399 .2876 1.2488 387) 670) 854) 360) 369) A51) A87) 867) 360) 859) 377) 418) 437) (117 887) 864) 864) 346) 464) 862) 3693 8503 849) 419) A3A) 840) 8573 64.9(314.6 13,7 11,379 5508 994.499 3309 NE 11P 48 BLY 12.439 1768 10.366 324(1571 158'0 PING AIR 11,375 195.6 99,539 292,725 292.725 86.560 10,857 11.077 11.934 12.334 12,330 169,231 11.935 257,574 235,674 12.341 15.431

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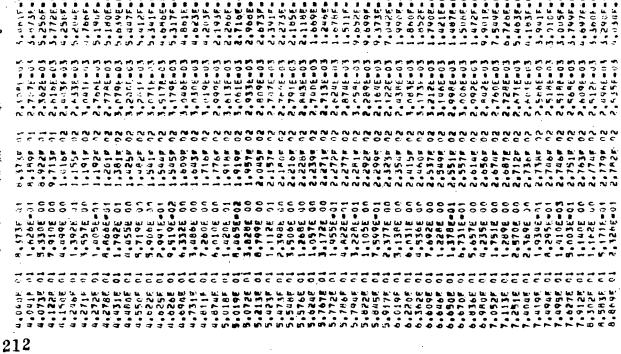
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LIAC 0.25 0.04 2446 54,133 170,4 0,21 0,03 0.01 0.04 19/4 16.013 136.6 0.56 0.00 0.01 <u>-</u> 2432 52.141 169.4 0.21 c1/7 34.359 150.9 0.36 0.36 0.36 0.21 IVAC 2357 50.603 164.2 2178 34,455 151.0 4393 52.167 166.7 6217 41.509 154.4 2167 38.811 150.2 2017 25.035 139.8 1967 19,061 137.8 5.577 200.9 0.010 £15.6 2446 55.224 171.5 2406 50.09/ 17c.y 2406 12.190 172.9 6.064 200.7 1.475 200.7 3 2002 5116 3116 ¥/05 ¥ 26.718 2719 26.718 2248 1.789 4022 2.102 0.0608/ 14.553 0.0848 2659 0.702 1802 2.200 0.59315 14.424 0.0444 2698 897 7.045 6365 1.837 U.V6130 15.531 0.9892 2672 0.382 1022 2.144 0.06150 15.551 0.9892 28,909 2656 28,908 1815 2,775 5035 1,908 0,70569 14,261 0,0789 26.408 2655 26.408 2618 0.467 1223 1.985 0.64154 14.201 0.0868 2797 2160 2,179 4707 2,086 0,71280 14,353 0,0786 2718 1986 2.416 4808 2.059 0.70742 14.353 0.0792 2703 2018 2.304 4649 2.061 0.70038 14.353 0.0800 2316 1,669 3867 2,197 0,65656 14,424 0,0858 2728 2281 1.684 3861 2.188 0.65546 14.424 0.0858 2724 2290 1.670 3824 2.188 0.65313 14.424 0.0862 7.232 6376 1.037 0.05624 14.201 0.9892 2696.0 0.0068 0.0784 2565 0.956 2454 2.216 0.66403 14.424 0.0932 2682 2566 0.801 2055 2.206 0.59687 14.424 0.0945 DA/A 926 2.144 U.U5629 14.201 2.805 5105 1.908 0.04154 14.261 2.508 4905 2.047 0.71018 14.353 4 / v ر ۲. MACH 2697 2676 U.\$46 2698 882 30v V 20.054 2725 26.909 2656 28.900 1762 2719 2750 6697 2728 24.909 25,195 706.82 707.407 26.408 20.024 25.136 25.128 25.127 25.229 25.141 1.2842 28.909 1.39/5 28.908 20.060 MULT 1.2827 1.3070 1.2842 1.2431 1.2875 1.2875 1,3295 1.5019 1.5000 1.3051 1.2831 1.2875 1,2934 1.2474 1,3323 1.5208 1,2957 1,3452 1,2925 294) 418) 754) 883) H83) A83) (c 483) 850) 349) 850) 335) 850) 821) 407) (915 878) 422) 865) 437) A81) 562) 890) 471) 575) 867) 5773 874) 155) 856) 753) (104 H6 %) 951) 884) 161.10 30.26 /61.1(26.6(16.7(122.11 424.2(424.9(695.1(574.6(-51.4(143.9(61.16 140.70 51.16 710.20 20.06 219.86 26.6(205.7(960.7(37.4(,20.6(364.86 295.7(95.30 22.30 21.2(34.6 10.400 500.721 3185 10.400 10.599 1415 NLET UPNHSK 99,424 3185 2045 4067 355 522R \$105 994.749 3296 32.40 3245 1897 300.721 5165 9.145 1361 11.562 10.353 0.172 231.010 20.249 0.153 601.006 11.562 175.543 195.558 108.272 38.3AB 65.908 178.089 15.783 67.294 44,123 12.399 93.572 16.502 19.453 20.481 94.812 20.492 90.61/ 0.000 0. SPIKE 11P NS NEET DANKSK AIND TUNNEL INC TONE UMBIISTUR BUSTOR PUMBUSTUR HUSTOR OMEUSTOR OPBUSTOR CUMBUSTUR OMBUS TUR . 460 2.716 105.57 225 225 0.00.0

READING # 0089" BLOCK = 183 TIME = 552.475 BACH 7.2 Pl = 494.149 11 = 5246.3

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##### 5000 MALH ###################################	19 1 2 21
	14 14 21 2.55.2 567.5 (40) 2.60.13 5.1 (40) 2.61.1 561.7 (67) 2.61.1 561.7 (67) 2.61.1 561.7 (67) 2.70.2 57.6 (68)

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MEADING # 0089 BLUCK # 163 | TIME # 552.475

Y	, ,	1 3 4 1 /		0 = U / / 0 •	1.50.00		2444	0-1600	9446-0	0 4 4 5 F + 0	.200f -0	0-3660.	. / sof = 0	.>v/E=0	.653E=0	0-1001-		. 23.5£*0	.1<11-0	• \$45E=0	0.4016.00	- 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1	. UVUE-0	.045t =0	. 584t = 0	.605E=0	.545E=0	.105E-0	. 2025 - 0	.556E-0	.8/7E=0	1.5566-02	0.4046	.657E-0	.0016-0	.656E=0		. / U DE - U	.806E-0	.304E-0	.108E=0	0-3/18.	0.0000	7856	-257E-	.40/E-	.435E-	.bute	· ponq.
CF	. I I I .	1000		7486	17.14		20.00	9592-0	.931E-U	.218E-U	. 300E .	.375t-U	.757k-u	.316E-U	0-3670.	11636-0	1135-0	-205-0	.957E=U	.9375-0	8056	9466-0	4956t = 0	.985£-U	. 809E = 0	976E-U	.102E-U	.410E=0	7546	636E-U	.568£*U	5.1332-03	4777	.5ult-0	.513£-0	.503k=0	- 3 / 1 C - 0	.162t-U	.130£-0	.070E-0	.000E-0	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		976E=0	9886-0	.924k-0	.8/5t-v	.908E-0	.908t-u
CIPAG	0 30 40	1000	0 300	C 2440	C 4170	2 1/1/1	4423E U	. 3527 ·	.234E 0	0 360H.	.450E 0	. 501E U	6 345¢	. 360E	0 35 C 4		- 7257.	1 3/8/	. 41 KE 0	0.3254.	150F	.176E 0	6 3017.	. 446E 0	. 272F 0	.274E 0	.279E 0	0 3856.0	3045	.332E 0	.367E 0	2.438E 02	3775	.583E 0	.584E 0	0 HOVE.	O HOPE O	. 707E 0	.721E 0	.749E 0	• 775E 0	0.444	7.865	191E 0	.603E 0	. AlbE 0	.021E 0	. 624E 0	• 824E ·
UDHAL	4		1776	100	568		.503E	.423t-	.966L-U	0 355t.	.121E U	.15hE U	.7/9E v	3 - 74 C - 6	0 3/1/	. 435C C	030E U	.041E U	. 290E	. 564 104 104 104	1245	.577£ U	. 8csE U	0 3656 0	3966	.076E=0	.270E-U	. 5466-0	819E=0	. 9186	.562E 0	7.097E 00	1551	.163E U	.172E=0	0 1 1 0 1 6 ° ° ° ° ° ° ° ° ° ° ° ° ° ° ° ° ° °	4056 0	.847E U	.417E U	AOSE C	95/25 0	010001	TRUE	247E-U	-210E	.250E	58	.360E	3
*	10405	40.40	7 7 7	1236	150E	24.5E	471E	. 27.2t	.278E	431E	4 H O E	.550E	11.00	. O C 2 C	1000	7 7 1 5	811E	. 47 4E.	0196	27.00	4275	.473E	5486	19/64	.767E	.772E	. 7 H 6 E	- / c	. 445E	.917E. U	019E 0	6.220£ 01	60AE O	. 646E 0	0 3000	0 C C C C C C C C C C C C C C C C C C C	.980E 0	.052E U	113E 0	. 251E 0	4000	97.53	176t	.627E	.912E	.302£	.583E	3698	10 / D.

PEAUTING # 0089 PLOCK # 183 TIME # 352,473 MAUR 7.2 PT # 994,749 TT # 5296.5

HAMJET PEHFÜRMANCE

ENGINE PERFORMANCE		Intel	
CALCULATED THRUST	329. (18F) 1234. (LBF) 786. (LBF=SEC/LBM) 2946. (LEF=SEC/LBM) 0.2503	Y EFFICIENCY SUPERSONIC	0.000 (UEGREES) 0.0000 0.0000 0.0987 0.0076 (PSI) 0.0049
SIFEAM IMMUST	PERFURMANCE 5397. (LUF) 552. (LUF) 1270. (LUF-SEC/LHM) 0,3721	PROCESS EFFICIENCY = SUBSONIC IC ENERGY EFFICIENCY = SUBSONIC IC ENERGY EFFICIENCY = SUBSONIC IC ENERGY EFFICIENCY = SUBSONIC IPY AT PO = SUBSONIC	0.9119 0.92119 0.92119 -22.20 12.40 (610/[87]
HUMESTUR AND FURLES		RUNBUSICA	
INLET FRICTION CRAG. INLET MUMENTUM CHANGE. CUMBUSTUR FRICTION URAG. COMBUSTUR MUMENTUM CHANGE. NOZZLE FRICTION DRAG. NOZZLE FRICTION DRAG. NOZZLE FRICTION DRAG. NOZZLE PRESSURE INTEGRAL. EXTERNAL PRESSURE INTEGRAL. EXTERNAL PRESSURE INTEGRAL. CANTAY FORCE. CALCULATED LOAD CELL FONCE. MEASURE LOAD CELL FONCE. CALCULATED LOAD CELL FONCE. MEASURE LOAD CELL FONCE.	86.4 (LbF) 171.9 2 (LbF) -24.96 (LbF) -24.04 (LbF) -24.04 (LbF) 726. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 750. (LbF) 7777. (LbF) -695. (LbF) -695. (LbF) -695. (LbF) -695. (LbF) -695. (LbF)	FULL—AIM MATID	0.0000 0.0000 0.000000
STATIONS		FUEL INJECTORS	
NUMINAL CUML LEADING EUGE SPIKE THANSLATION INLET THRUAT CONL LEADING EDGE CONL LEADING EDGE NUZZLE SHMOUD THAILING EDGE SIRUI LEADING EUGE STRUI LEADING EUGE	10.7210 (IN) 10.7210 (IN) 40.4010 (IN) 14.945 (IN) 14.945 (IN) 17.325 (IN)	INJECTURS STATION VALVE 1A 40.400 A 47.100 1B 47.706 B 44.300 2A 40.250 3A 55.471 3H 46.206 C	w .

Reading 89

t = 316.47 sec.

Recomputations with surface pressure substitutions.

READING = 0089 BLOCK = 143 TIME = 316,473 MACH 7.3 PT = 995,499 TT = 2722,9
RAMJET PERFORMANCE

•		9	77.	**															
	ETAC	3	amed 12, 206,276				·		0.07	0.33			- (90. 0	80.0	0.02	0.29 0.00
	PHI		3						0.18	0.18			01.0	01.0			0.29	0.29	0.29
	IVAC	-	185,1	180.2	180.2	155.2	156.4	156.4	154.3	153.7	7.151		143.4	147.5	137.4	136.7	126.4	124.3	122.8
	æ	5,798	0,921	5,981	986.0	58,282	53,669	12.857	56.845	54,408	55.911	700	127.65	40 to 10 to 60 to 10 to	41,279	2281 40,797	25,701	21,968	19,704 122.8
	MONTK M	2980	3059	3075	3075	2565	2584	2584	2564	2555				5000 P	2292		2109	2073	2048
	A/AC	0.9901	0.9901	0.9901	0.9901	0.0789	0.0867	0.0867	0.0789	0.0786	0.0792			0.0859	0.0858	0.0861	0.0932	0.0944	6+60•0
	*	16.528	16.528	17.059	17.059	16.528	16.528	16.528	16.620	16.620	16.620	16.630	70.020	16.679	16.679	16.679	16.679	16.679	
0 R T	A/X	0.06518	0.06518	0.06727	0.06727	0.81836	0.74396	0.74396	0.82283	0.82585	0.81959			0.75841 16.679	0.75876 16.679	0.75645	0.69887	0.68978	0.68609 16.679
REP	ഗ	1.779	2.087	1.779	2.087	1,843	1.843	1.922	1.972	2.017						2,086	2,113	2,102 0	2.097
~	VEL	5724	606	5721	943	4583	4642	1112	4445	4239	4390	4260	44 E	3504	3501	3470	2366	2049	1648
₹	MACH	7.302	0.372	7,251-5721	0.386	2,854	2,947	0.466	2,506	2,116			761	1.685	1.683	1.668	1.024	0,889	0.801
S	SONV	2469 784	2469 2445	2469 789	2469 2444	2424 1606	2424 1575	2424	2498 1774	2587 2003	2480 1767			2495	2494	2489 2081	2486 2311	2439	2416 2305
	VOLWT	28.909	28.908 28.908	28,909 28,908	28.908 28.908	28,909 28,908	28.909 28.908	28.908 28.908	26.979 26.979	27.244	26.959 26.959	26.913	27.091	25.861 25.861	25.859 25.859	25.850 25.850	25.911 25.911	25.820 25.820	25.798 25.797
	GANMA	1.3019	1,3019	1,3019	1,3019	1,3053	1,3053	1.3053	1,3083	1.2974	1,3099	1,3122	1.3056	1,3140	1,3141	1,3146	1,3138	1,3186	1,3205
		714)	714)	714)	714)	684) 266)	684)	684) 659)	721) 328)	792)	709)	693)	733)	7117	710)	707) 467)	707) 595)	673) 589)	658) 590)
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	-	2723 256	2723 2667	2723 259	4 2723 2663	2618 1092	2618 1048	2618 2534	2588 1251	2826 1640	10 2546 1239	2494 1262	12 2630 1756	13 2465 1676	14 2462 1675	2450 1677	2452	2342	2293 2076
	2	5,499	1,175 0,223	5,499	11,175 10,152	32,041 10,420	.980	15K .04.177 90.605	39.443	71.741	216,454 12,426	204.997		03,023	02.911 21.698	101.169 21.850			
		2	0.600 0.600	Ξ :	0.600 0.600	40.400 332.041 40.400 10.420	40.400 332 40.400 332	10LE1 DNNRSK 40.400 104 40.400 90	40.410 2	40,733 1 40,733 1	CCMBUSTOR 41.223 2	COMBUSTOR 41.500 2	COMBUSTOR 42.460 1	ē	COMBUSTOR 42.718 1	42.783 142.783	44.310 44.310	5 6	•

ETAC	0.01	0.04		•	0.07		0.27	9. ¢	n + 0	N	0.33	0,40	t .	. =			0.38	•	1
PHI	0.29	0,59	59	59	.59	5.0	59	Q.	0.82	0.82		0.83	**	83	8	.83	כיו	n	,
IVAC	25.1	22 e B	2.8	60	32.7	4.04	5		89	67.3	71.5	176.5 (,		6	6.48	87.2	7.2	
	521 1	975 1		119		-	• -		. 09	120 10	81 1	13	19	45 1	1 12	47 1	98	96 1	
•	18.	17.	17.	17.	18.	19.	14.	23.	23.8	27.	23.8	20.1	19.3	18.	18.	±.	14.8	13	
MOM	2086	2066	2068	2168	2233	2397	2549	2793	2794	2837	2908	2999	3019	. .	3057	3141	3180	3180	
A/AC	9260.0	0.0979	0.0979	0.1029	0.1056	0.1149	.12	0.1552	0.1554	0,1662	0.1950	0.2379	0.2480	.263	26	340	0.3679	0.3690	ļ
3	16.679	16.833	16.833	16.833	16.833	16.833	6.83	16.954	16.954	16.954	16.954	16.989	16.989	•	- 98	•	16.989	16.989	,
#/#	0.66735	0.67162	0.67132	0.63896	0.62225	0.57190	52	0.42650	.0.42594	0.39836	0.33951	0.27884	0.26745	0.25211	.2469	0.19504	0.18030	916110	!
s	2.094	2,268	2,254	2,245	2,270	2,317	.321	2,515	2,514	2,461	2.479	2,514	2,521	,532	.536	.549	2,500	2,567	
VEL	1786	1722	1722	1724	1900	2164	8	3590	3605	4381	4526	1494	4658	4682	4692	4931	5314	4938	
KACH	0.775	0.707	0.720	0.727	777.0	0.827	্	1.276	1,283	1.815	1,857	1.845	. 833	1,815	.81	1.928	2,372	1.873	. (
SONS	2408	2529	2486	2467	2555	2740		3091	3089 2810	2928	2970	3041	3060			3114	3003	3158	3149
	25.809	23,380	23.301	23.289	23.467	23.902	974	22.981	22.977	22.541	22.673	22.813	22.877	22.972	23.006	23.071	22.752 22.755	23,249	23,216
GAWWA.	1,3209	1.3239	1,3281	1,3295	1,3206	1.2999		1.2666	1,2668	1,2899	1,2839	1.2737	1,2705	1,2656	1.2639	1.2602	1.2784	1,2502	1,2523
	6531	716)	686) 627)	673)	736)	884)	902) 840)	1205) 956)	952)	652)	(074)	(151)	1173)	12051	1216)	(239) 765)	1112) 554)	(1298) (828)	1286)
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_ =				22 1144 970	2332 2332 2124	24 2777 2515	25 2832 2656				3133	3331	3391 2270	3479 2355	33 3510 2384			3731 2508	3696
	.927 .432			67.421 48.059		65.752 42.958									0 48.186 8.426			38,345	
NO.		5	<u> </u>	Ž (œ q	ž		z . c	<u> </u>	z . 9	<u> </u>	<u> </u>	<u> </u>	x (<u>«</u>	6.243 6.243	<u> </u>		

	ETAC	0.32	0.26	0.25	0.20	. 99•0	9.8	0.86	06.0	•	36.0	0.95	0.95	. 95	0.95	1.00	0.95
	PHI	0.83	0.83	0.83	0.83	0.83	0.83	0.83	0.83	83	0.83	0.83	0.83	0.83	0.83	0.83	0.83
	IVAC	187.4	187.5	.87.7	87.9	188.5	188.6	188.4	188.2	~	6.06	243.9	262.8	250.7	271.6	285.4	228.6
٠	_	339	720	811 1	919 1	816	. 877	980	.036 1		376 1	3,804.2	.432 2	893	.404	. 263	3,425 2
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	KOMTK	3183	3186	3188	3192	320	3203	3201	3198	3197	3244	4144	4465	4258	461	4848	3883
-	A/AC	0.3675	0.3687	0.3693	0.3753	0.3777	0.3650	0.3553	0.3749	0.4032	0.4032	1,9371	5.7738	1.9371	6.0782	2,8325	1.9371
6.	3	16.989	16.989	16.989	16.989	16.989	16.989	16,989	16.989	16.989	16.989	16.989	16.989	16,989	16.989	16.989	16.989
T = 2722.	W/A	0.18050	0.17990	0.17960	0.17675	0.17562	0.18173	0.18666	0.17693		0.16449	0.03424	0.01149	0.03424	0.01091	0.02342	0.03424
11 .66	s	2,458	2,401	2,382	2,305	2,597	2,619	2,618	2,623 (.631	.660	.631	631	2,660 (999	.423	2,672 (
6648.846	VEL	5468 2	5623 2	5665 2	5796 2	9694	205	130	7 4104	3858 2	3668 2	7149 2	8018 2	7316 2	8279 2	968 2	6437 8
PT ::	MACH	• 686 5	.154 5	.320 5	.078 5	.643 4	,358 H	.324 4	.276	214 3	.125 3	.827 7	.742 8	7 577.	.717 8	556 8	.358 6
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7	SONV	2929	1783	2825	2749	3247	3326	3333	3338	3346	3404	3346	3346	3404	3404	3455	3314
73 MACH.	POLKT	22.562 22.563	22.370	22.323	22.171 22.171	23.659	24.156	24.22 2 24.388	24.316	24.406	24.241	24.406	24.406	24.241	24.241	24.740	24.381
= 316,473	GAMMA	1,2884	1.2983	1,3008	1,3089	1,2237	1.1863	1,1822	1.1752	1,1664	1,1562	1.1664	1,1664	1,1562	1,1562	1,1859	1.1621
TIME :		1036)	958) 329)	938)	874) 204)	1437)	1660)	1617)	1639)	(668) (457)	1742)	1604) 797)	535)	1742) 885)	1742) 586)	1782) 283)	1636) 965)
143	Ι,	19.36 18.36	32 6 448.3(-183.5(447.5(34 6 445.2(-226.0(441.90	34.	429. 88.	419.5(1 97.6(1	39 4 417,9(16 120,5(1	0 3 554.6(1 285.7(1	000	42 5 417.9(1 -866.9(554.6() -515.1(ສຸທ	۰.6	394.2(433.8(
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9800 :	۵	3.	118 2,	137,32	263	32	29	29.	27		25,596 12,655	, 52°	D 1	י ע	25.596 0.156	in c	14°5
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READ1		57.9	50.224 50.224 50.224	588 88.	59.17	60.19 60.19	62.20	63.6%	66.02 66.02 66.02	COMBUST 66.463 66.463	66.46 66.46	98.66	88.65 88.65	99.66 88.66	989.66	999	88.66

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•	CF.	108F-0	7075	362F-0	616E-0	403E-0	633E-0	041E-0	778F-0	0.3070	200E-0	.201E-0	.174E-0	.566E-0	.267E-0	068F-0	.037E-Ü	.316E-0	.098E-0	1415-0	761E-0	.775E-0	.895E-0	.907E-0	869F-0	.835E-0	.656E-0	•973E-0	1092E-0	022F-0	.902E-0	.181E-0	.197E-0	. 436F-0	.521E-0	.611E-0	3.5095-03	403E-0	.341E-0	.316E-0	0.20.00	200F-0	1455-0	.145E-0	.181E-0	•206E=0	.124E-0	1305-0	0 - 3 o t •
	CURAG	3735 0	389E 0	.922E 0	.713E 0	.016E 0	.155F. 0	191E U	201F 0	3815 0	.425E 0	.482E 0	.536E 0	.538E 0	586F 0	.611E	.668E 0	.710E 0	.821E 0	876E 0	.009E 0	.171E 0	.207E 0	.259E 0	.294E 0	.340E 0	.342E U	.349E 0	371E 0	381E 0	4410E 0	• 448E 0	534E 0	.729E 0	.747E 0	.748E 0	2. 738E UZ	.889E 0	.911E 0	.927E 0	9875	.990E 0	000E 0	.000E 0	•006E 0	• 019E 0	00005	.040F	1
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	×	0 7	041	073	.122	.150	.246	27.5	278	431	.480	.550	622	625	169	731	.811	. 874	210	.072	.213	423	674	576	.624	.767	.772	, œ	822	.845	.917	019	200	609	9,49	650	6.836E	.980	052	.15	404	419	#6#	•495	627	30.5	583	.869	
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RAMUET PERFORMANCE

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	ICIENT: ECOVERY VERY - S	1 1 m ·	COMBUS	NOZZLE T COEFFICIENT CT	FUEL INJECTORS	STATION 40.400 42.708 44.300 50.43 46.25 55.473 57.658
	ANGLE OF ATTACK MASS FLOW RATIO			STREAM THRUST COEFFICIENT - CS COEFFICIENT - CT	FUE	
	E OF ATTAC FLOW RATI TIVE DRAG TING PRESS A PT2 L PRESSURE L PRESSURE	INLET PROCESS INLET PROCESS INLET PROCESS KINETIC ENERGY KINETIC ENERGY ENTHALPY AT PO ENTHALPY AT PO ENTHALPY AT PO	FUEL-AIR RATIO EQUIVALENCE RATIO. COMBUSTOR EFFICIEN COMBUSTOR EFFECTIVE INJECTOR DISCHARGE	S C		INJECTORS 18 10 20 20 30 38
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	903. (1024. (2083. (2083. (0.6077			29,57 -0.00 -0.00 -0.00 72,05 -603 -30,94 -468,		34.884 1.7230 40.400 36.607 74.900 57.863 66.463
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engine	IMPULSE MPULSE SEFFICIE	TIVE-COO	MOMENT GE. GE. GA. GA.	NA GENERAL STANDER OF THE GRAL STANDER OF THE GRAL STANDER OF THE GRAL STANDER OF THE GRAL STANDER OF THE GRAND	15	G EDGE
	CALCULATED THRUST	STREAM THRUST	INLET FRICTION DRAG	NOZZLE STRUT DRAG. NOZZLE STRUT DRAG. NOZZLE WOMENTUM CHANGE. NOZZLE PRESSURE INTEGRAL. EXTERNAL FRICTION DRAG. EXTERNAL PRESSURE INTEGRAL. TOTAL EXTERNAL DRAG. CAVITY FORCE. CALCULATED LOAD CELL FORCE. HEASURED LOAD CELL FORCE.		NOMINAL COWL LEADING EDGE
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Recomputations with surface pressure substitutions.

SUPMARY REPORT

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ETAC	Trail.	channel	(al-100) (52)			·.		0.22	0,40	90.0	0.01	0,20	0.05	0,05	*0°0	0.03	00.00	00.00
IVAC · PHI	~	3	7					0.17	0.17	0.17			0.28	0.28	0.28	0.28	0.28	0.28
IVAC	168.4	170.7	168.3	168		146.3	146.3	144.4	143,9	142.0	140.0.0.17	131.3	128,3	128.2		117.7	115.8	114.7
ø	5.888	1961	5,983	1.002	59,302	54,609	13,261	56,103	54.081	56.849	54.828	42.918	41.061	40.965	40.416	25,299	21,697	2081 18,952 114.7 0.28 0.00
30X	3027	3068	3076	3076		2630	2630	2610	2602	2567	2530	2373	2326	2324		2133	2100	2081
A/AC	h066°0	0.9904	4066.0	4066°0	0.0789	0.0868	0.0868	0.0789	0.0786	0.0792	0.80.0	0.0848	0.0858	0.0858	0.0862	0.0932	0.0943	6460*0
3		17.976								18.074.0.0792	18.074	18.074	18-133	18.133 (18.133	18.133	18.133	18.133
W/A	0.07087 17.976	0.07087	5344 1.743 0.07204 18.273	0.07204 18.273	2,853 4290 1,807 0,88956 17,976	0.80869 17.976	0.80869 17.976	0.89429 18.074	0.89759 18.074	2.447 4106 1.940 0.89082	2,378 4000 1,936 0,88195	0.83219	0.82542	0.82467	0,82111	0.75938	0.75038	0.745 1635 2.054 0.74568 18.133 0.0949
'n	5346 1.743 (2,052	1.743	2,052	1.807	1.807	1.886	2,210 4037 1,967	3877 1.995	1.940	1.936	2.000	2.048	2.048	2.048	2.062	2.058	2,054 (
VEL	5346	878		895	4290	4345	1055	4037		4106	4000	3318	3201	3196	3167	2144	1861	1635
MACH	7,340	0.381	7,314	0.388	2,853	2,948 4345	0.468 1055	2,210	1,957	2.447	2,378	1,653	1.600	1,598	1.584	0.989	0.852	0.745
SONV	2331 728	2331 2307	2331	2331 2306	2290 1504	2290	2290 2253	2425 1826	2489 1981	2353 1678	2329 1682	2393 2007	2370	2369	2363 2000	2328 2169	2303	2289 2196
MOLWI	28.909 28.908	28.908 28.908	28.909 28.908	28.908 28.908	28.908 28.908	28,908	28.908 28.908	27.174	27.356 27.356	27.018 27.017	26.968 26.968	27.158 27.158	25,994	25.992 25.992	25.981 25.981	25.950 25.950	25,924 25,924	25.919 25.919
GAMMA	1,3121	1.3122	1,3121	1.3122	1,3151	1,3151	1,3152	1.3111	1.3036	1,3185	1.3210	1,3136	1.3220	1.3220	1,3226	1,3254	1,3276	1,3287
	623) 53)	623) 608)	623) 53)	623) 607)	598) 230)	598) 221)	598) 575)	676) 351)	724)	627)	611) 291)	655) 435)	631)	631) 427)	627) 427)	605) 513)	589)	581) 528)
± ,	495.0(-76.2(95.	35.	0 495. 479.	0 469. 101.	469. 91.	469.0(478.8(153.1(476.4(35	470.4(150.6(465.2(260.4(465.1(260.9(464.5(448.9(357.0(374.9(437.36
,	2408 221	2408 2355	2408 223	4 2408 2353	2318 951	2318 913	2318 2240	2451 1343	26 <u>1</u> 4 1609	2282 1115	2227 7111	2381 1640	2222 1551	2220 1551	2206 1549	2136 1838	2084 1859	2057 1883
م.	.999	11,162	999	162	.827 .547	982	.783 .778	187,486 16,915							97.655 23.882			
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	ETAC	00.00	0.04	0.01	00.00	0.02	0.17	0.21	0.35	0 35	0.24	0.29	0.36	0.37	0.39	0.40	19 17 0	0.73	0.74	0.75
	PHI	0.28	0.56	0.56	0.56	0.56	0.56	0.56	0.79	0.79	0.79	0.79	0.80	0.80	0.80	0.80	0.80	0.80	0.80	0.80
	IVAC	117.0	114.8	114.9	120.1	123,4	131.9	140.2	152.5	152.6	155.0	159.1	164.2	165.4	167.0	167.6	172.6	176.2	176.4	176.8
	G	106.	349	338	474	464	505	. 644	.809	.871	.065	787	.910	.375	395	. 064	. 997	550	349	11,199
		71. 23	71 00	02 17	97 16	57 17	3.18	7	09 23	0 23	. 27	31 23	19	2 19	2 18	93 18	5 14	52 11	5 11,	
	MOM	212	210	210	219	225	2413	2565	280	281	2855	293	3031	3052	3082	309	318	325	3255	3262
	A/AC	9260•0	0.0979	0.0979	0.1029	0.1057	0.1150	0.1263	0.1552	0.1554	0.1662	0.1950	0.2379.	0.2480	0.2631	0.2687	0.3401	0.3678	0.3691	0.3716
Ψ.	3	18.133	18.294	18.294	18.294	18.294	18.294	18.294	18.421	18.421	18.421	18.421	18.456	18.456	18.456	18.456	18.456	18.456	18.456	18.456
= 2408	W/A	2515	.72982	.72937	.69436	.67590	.62132	.56555	.46340	.46280	.43283	36889	.30292	*29054	27388	.26820	.21188	19591	.19525	19392
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μ	MACH	0.726	0.657	0.670	0.674	0.732	0.788	0.667	1.257	1,264	1.768	1.792	1.743	1.776	1.766	1.763	1,862	1.271	1.244	1.230
7.3	SONV	2277 2188	2407	2363 2282	2345 2264	2362 2267	2542	2580 2499	2903 2630	2902 2626	2759	2810 2315	2901 2427	2905 2416	2931	2940	2971	3219 2986	3227	3234
3 MACH	MOLWT	25.918 25.918	23.567 23.567	23.491	23.480	23.519 23.519	23.878	23.977	22.906	22.903	22,579 22,579	22.714	22.881 22.882	22.898 22.899	22.977 22.979	23.005	23.098	24.043	24.086 24.143	24.127 24.188
327,27	GAMMA		1.3311	1.3352	1.3366	1.3348	1.3167	1.3124	1.2876	1.2878	1,3045	1,2983	1.2871	1,2865	1,2828	1,2815	1.2768	1.2202	1,2171	1,2142
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3 MACH	WOL NT	24.135	24.225		24.374		24.585		24.710		24.659		24.521		24.388			24.508		24.328		24.417		24.4.17		24.328			24.548		24.958		24.416	1
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READING # 0490 PLOCK # 92 TIPE # 206.263 AACH 7.5 PL. # 995.494 TE # 23946

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HEADING # 6.190

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REAUTING & ARON " FLACK # 119 TIPE # 25". 523 - 4664 7.3 PT # 995.244 TT # 2021.5

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10.972 3636 508.5(1692) 1.2575 17.120 3644 0.155 1323 ***45.3(556) 1.3575 17.134 2281 3.330 7598 3.414 0.000944 7.470 2.4626 0.155 1323 ***45.3(556) 1.3559 17.134 2281 3.330 7598 3.414 0.000944 7.470 2.4620 0.155 145 4466 260.2(210) 1.2438 18.064 3607 0.155 62841621.9(242) 1.1936 18.782 1406 5.938 9736 2.919 0.02930 7.470 0.9955 0.155 6285 448.2(1323) 1.2355 25.608 3180 0.156 1368 1794 3.652.1 5.455 1.450 3.404 2034 2.41 0.00 2.477 0.004 2.410 0.005	372Z0	DO REGER		17		•		•		•		•	•			,	,
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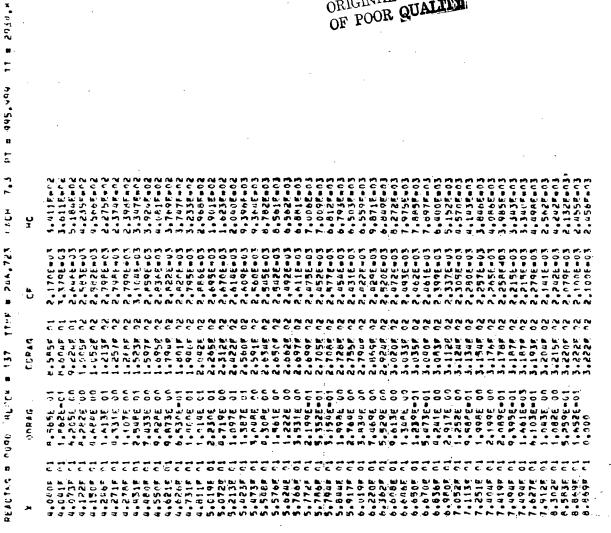
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Reading 91

t = 175.65 sec.

Injected fuel appeared to have not ignited.

During the first fuel schedule (see fig. 6 (d)) the injected fuel appeared to ignite at time 179 seconds.

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		32	700	26.955		7.260	6120	1.814	0.05953	13,781	0.9039	2657	5.661	192.8		
0 h	94	15) 1	2903	28.955	2608	0.358	927	2,122	0.05953	13,781	6£06°0.	6788	0.857	202.3		
. 3 % .		33.	3967	28.956	2608	7.167	6113	1.814	0.06306	14.599	0.0039	2.832	5.440	192.6		
0 4	•••	1 (90	2962.	28,955.	2608	0.384	991	2,122	0.06306	14.599	0.4039	2815	0.971	192.6		
•		200	2944	28,955	1826	2,537	4632	1.908	0.67881	13.781	4010.0	2212	298-97	100.5		
•	# 7 6 7 1	1 (08	3548	28.955	1792	2.628	4710	1.908	0.01710	13,781	0.0672	2253	45.174	162.1	•	
	, 1	66	2976	28,955	2560	0.487	1227	1.968	0.61710	13,781	0.0672	2233	11.767	102.1		
, 55 c	4.1. 5.1.7.	800	.2944	28.955	1956	2,537	4632	1.908	0.67872	13,781	0.0793	2212	46.855	160.5		
.57. .:2.	-	55	.3947	26.955	2557	2,532	1298	1.907	0.08117	13.701	0.0790	2208	48.91	3.001		
	6.20	143	3497	28.955	1945	2,467	4552	1.911	0.67639	13,781	0.0795	2187	47.651	156.7		
, , , , , , , , , , , , , , , , , , ,		55	2953	28,955	2549		0477	•	0.66936	13,781	0.0	2166	46.57	151		
, 56,		2.0	3425	26.955	2539	2,234	5627	1:929	6.63193	13,781	0.0851	2118	42.162	183.4		
	3.4 m	533 1	.2962	28,955	2537	2.207	4261	1.931	0.62453	13,781	0.0862	2105	41,361	152		
. 25.	• •	1 (24)	2963	28,955	2536	2.200	4252	1.932	0.62223	13.781	0.0865	2102	41.117	182.5		
	49	110	3373	28,955	2823	2.044	4050	1.945	0.57528	13,781	0.0935	2045	36.207	7.84		
200	*	500	3358	28,955	2519	1,989	3978	1.948	0.56885	13,781	9760.0	2026	35,165	147.0		
. 95. 20. 51.	. K 3	169	3344	28,955	2514	1,932	3897	1.951	0.56558	13,761	0.0951	2008	34.257	1 45		
25 21 21 22	5.9(7/	42) 1	3346	26,955	2508	1,924	3877	1.952	0.55049	13,781	n.0977	1961	33,165	144.9		

11	10 1
### TIME # 166,451 HACH 7,3 PT # 995,499 TT # 1069,0	10 12 13 14 15 12 15 15 15 15 15 15
### 1266.451 HACH 7.3 PI # 995.499 TI # 3069 ###################################	0.00CK m 86 TIME m 166.451 MACH 7.3 PT m 995.499 TT m 3069 1 1 2 2
## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH VEL ## GAMMA MULMT BONV MACH MACH IN ## GAMMA MULMT BONV MACH MACH IN ## GAMMA MULMT BONV MACH MACH IN ## GAMMA MULMT BONV MACH MACH IN ## GAMMA MULMT BONV MACH MACH IN ## GAMMA MULMT BONV MACH IN ## GAMMA MULM BONV MACH IN ## GA	BLOCK # 86 TIME # 186,451 HACH 7.3 PT # 995,49 10 19 12
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i	•					GAMAD	GAMMA MOLWI SONV MACH VEL S	SONV	HACH	V.		* / Y		1/10	A/AC . HOFTE	G	1 v A C	IVAC PHI ET	7
33 <u>2</u>	60.175 60.175	66.249	2000	364.6 564.6	692)	1.3040	28,955	2035	6	9	i	(692) 1,3040 28,955 2435					•		
្ត	MBU. TOR			32 4	(»C y	1 • 2 • 2	K0.03	12/5		000	953.	0.14640	13.781	0.5777	K113	2173 10.380 157.7	157.7		
4	*101				691)	1,3041	28,955	2433				691) 1,3041 28,955 2433							
٠	• 185				2573	1.3749	26.955	1582	2,950	9997	1.956	0.14742	13.781	0.3650	2166	2166 10.689 157.1	157.1		
ថ	HEUS TOR			33. 4										,	·	•			
•	.604			562,7((06.9	1,3042	28.955	2452				690) 1,3042 28,955 2452							
ō	• 604			130.9(360)	1.3742	28,955	1589	2,925	8797	1.955	0.15141	13.781	1555	0415	MARCHES TO SERVICE	196.4		
ដ	ABUB10R			70		, -	,			•		•	•		•				
3	.069			561.50	684)	1. 3043	28.955	2431				- 689) 1.3043 28.655 2431							
š	•069			135.20	264)	1.3731	28,955	1601	2.884	4618	1.962	0.14352	11.741	0.1749	2161	2161 10. 401 186	154		
ដ	*BUSTOR			35	•	•								7 - 7			1 0001		
š	\$44°				689	1.3044	28,955	2430						689) 1.3044 28.955 2430					
3	577.				265)	1.3730	28.955	1603	2.879	7 1 97	1.967	0.11144	11.78.1	0.4035	0416	644			
ž	ZZLE AL	قين	7		•) 												
ě	.681	55.779 B	2637		(694)	1.3044	28.955	2430				689) 1,3044 28,955 2430							
ĕ	.661	0.158	295		(32)	1.3981	28.955	1.162	4.936	5270	1.967	0.02778	11.781	1.0171	7110	3.941.340.c AFF	3.041		
ž	ZZLF P	9					•	•											
ě	.681	55,779			684)	1.3044	28.955	2450				689) 1,3044 28,955 2430							
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ž	677	214.202			694	1.3044	28.955	2410				689) 1.3044 28.955 2410							
3	. 444	0.154			610	1.3985	26.955	550	5.730	5475	1.876	0.04142	18.781	0104	2404	643			
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•	189	310.489		555.7	683)	1.3050	28,955	2422											
ŏ		0.073			(99	1.3956	28.985	811	6,862	5955	1.847	0.02778	13,781	66) 1.3956 28.955 811 6.862 5565 1.847 0.02778 13.781 1.9171 2420 2.402 175.6	2420	2.40%	175.6		

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CALCULATED THRUST	#253, (LBF) #253, (LBF) #253, (LBF#8EC/LBH) #1747	ANGLE OF ATTACK MASS PLOW MATTO ADGITTYE DRAG C LIVITING PRESSU DELTA PTESSU	ANGLE OF ATTACK	W 4 1 4 6 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	(DEGREES)	
BIREAM THRUST	PERFORMANCE 0. (LBF) 0.0000	INTEL PROBLEMENT PROBLEMENT PROCESSION PROCE	1014L PRESSURE RECOVERY = SUBSCRIC INLE! PROCESS EFFICIENCY = SUBSCRIC INLE! PROCESS EFFICIENCY = SUBSCRIC INLE! ENERGY EFFICIENCY = SUBSCRIC KINETIC ENERGY EFFICIENCY = SUBSCRIC ENIHALPY AT PO = SUPERSCRIC ENIHALPY AT PO = SUPSCRIC	40000000000000000000000000000000000000	(BTU/LBM)	
SOUND OVER TOPICS			CUYBUSTCR	•		
	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	FUEL BAIR RATIO COTBUSTOR RATIO COTBUSTOR ENTRA TOTAL COTBUSTOR ENTRA INTECTOR DISCRE	FUEL AIR RATIO	0007M		•
	•	VACULE GAREAL TIRUSA NOZALE COEFFICIENA & PROCESS EFFICIENA & NAMETIC ENERGY EFFICI	STREAT THRUSH COEFFICIENT & CO	1.0050		
BTATIONS		. :	FUEL INJECTORS			
NOHINAL COME LEADING EDGE NIET TANDLATION. INLET THROAT. TOWEL LEADING EDGE NOZZLE WHOUD TRAILING EDGE NOZZLE WHOUD TRAILING EDGE STRUIT LEADING EDGE GOFBUSTON EXIT.	12 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	12.46.1.46.1.46.1.46.1.46.1.46.1.46.1.46	6 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	₩ > - - - - - - -		

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00 995.749 3	69	616)	1.2906	28.956	6092											
100 0.166	• •	2	1.3967	28.455		7.170	6116	1.815	0.06288	14.563	2000.0	2807	5.977	1.92.7		
11.187 3	, 6	-	1.2902	26.955												
10, 10,183 3	071	797)	1.2923	26.455	2583	0.384	166	2,123	0.06288	14,563	2006.0	2807	0.966	196.7		
T THROAT			•													
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100 173-110 2		793)	1.2930	20.955	2578											
	, , ,	80	1.3457		1887	2,414 (4556	1.927	0.61669	13.780	0.0872	5500	43.663	154.7		
2 856.49. 001	007.	•	1.2930	28.95	2578											
100 73.709.2	.34	761)	1.2964	20,955	2534	905.0	1287	1.974	0.61669	13,780	5780.0	2200	12,330	159.7		
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13 21.957	360	604)	1.3203	26.373	2323	1.710	3973	2,142	0.68655	13,897	0.070.0	2161	42.390	155.5	0.26 0	25.
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100 95.174	670.	1	1.3006		2661											
100 21.,721	399.	562)	1,3295	26.086	2253	1,636	3686	2.128	0.67461	13.897	9080.0	2040	38.644	146.8	0.26 0	80
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98 55.624	2	656)	1,3063		2722											
108 23.872	707	0	1.3254	23.606	2471	1.200	2965	2,311 (0.63571	14,011	0.0861	1011	29,296	129.2	0.51 0	*0
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163 50.108		825)	,3126	23.722	2685											
163 23,737	9 8 67	S	1,3293	23.722	2434	1.200	1268	2.300 (0.63278	14.011	0.0865	1798	20.722	128.3	0.51 0	00.
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000 42.418	929	1159)	.2620	24,745	3027											
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187 41.205	650	S	1.3070	23.869	2724	٠										
163 35.258	616.	825)	1.5163	23,869	2477	267.0	1318	2.336 (0.57499	14.011	0.0951	1453	11.774	104.3	0.51.0	90.0
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	- -		- ^		CANA	HOLET	> X O S	I V E	. VEL	Ø	4 /2	£	A/AC	EOK 1	G	IVAC	E I	ETAC
'U IU	50.00		651	836)	1,3151	22,812 22,612	2710	0.583	1541	20402	0.56155	14.067	0.0978	1413	13,451	100.5	0.63.0	20.
02 0 26.632 2505 29.437 2373	2000		~	7703	1.3177	22.763	2685 2618	0.590	1545	2,395	0.56084	14.067	0.0979	1010	13.468		. 64.0	00
reg reg	1 2 K	•	651. 601.	814)	1.3181	22.756	2681	0.4.0	1568	2.395	0.55900	14.067	2860.0	101	13,621	100.5	0.63.0	00
	4 6 6 6	00	. 565 . 565	808)	1.3187	22.755	2673	0.797	2034	804.2	0.51817	14.007	0.1060	1443	16.382	102.6	0.63 0	00
led tel	led tel	3.9	554	11113	1.2625	23.805	3074	0.701	2096	2.515	0.47629	14.067	0.1153	1440	15.518	105.9	0.63.0	39
26.436 26 16.625 23		30	689	769)	1.3113	22,906	2738	0.822	2145	2.432	0.43488	14.067	0.1263	1540	14.500	109.5	0.63.0	•
		200	653	652)	1,3186	22,777	2667	1.138	2772	2.429	0.35340	14.067	0.1554	1640	15.27	116.6	0.63.0	.01
		E 4	E 6 6	7963	1,3198-	22,756	2655	1.260	3036	2.436	0.33052	14.067		1662	15.592	110.2	0.63 0	00
10	10	222	628 419.	791)	1.3203	22,755	2649	1.393	3234	2.447	0.28170	14.067	0.1950	1703	14,150	121.1	0.63.0	00
		6 T R	21. 622. 413.	767)	1.3424	22.654	2645	1.396	3232	2.457	0.23161	14.111	0.2379	1754	11,633	184.3	0.040	
		4.0	3792	544)	1.3213	22.649	2641	1.543	3476	2.471	0.22214	14.111	0.2480	1764	11,999	125.0	0 74.0	30
	-	8 A	519	550)	1.3215	22.649	2639	1.509	3417.	2.467	0.20940	14.111	0.2631	1775	13.139	125.8	0 90.0	8
10.416 33 4.240 27		-63	24 2 619,	1110)	1.2741	23,515	2999	1.239	3393	2.584	0.20478	14.111	0.692.0	1760	10.797	126.2	0 40.0	55.
		7 2 3	65 618 372	586)	1.3149	22.774	2700	1.509	3507	2,505	0.16200	14,111	0.3401	1822	30 30 50	129.1	9 9 9	90
	~ ~	33 415 600	26 2 616. 332.	787)		22.667		1.729	765	508	1671.		367	3	92	8.08	9	
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	₩	4 12 0 0 4 14 15 15 15 15 15 15 15 15 15 15 15 15 15		784)	1,3211	22,661	2640	1.806	3869	2.513 (26671.0	14.111	0.5675	1843	9.015	130.6	0 49.0	00
	~	100	300	465)	1,3575	22.650	2633	1.890	3972	2,532 (0.14943	16.111	0.3687	1846	9.224	130.6	0 4 6 0	00

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00 0.15 E TIP NS	962 7	-57.2(72)	1,3965			7,257	6145	1.817	0.05918	13.696	0.9036	2652	5.652	193.6		
11.17	5 5092	97.5	822)	1.2896	28,455	2616	•		:	į	-			:	:		
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11.17	2608 5	97.5	~	1.2896	28.95								-				
09.0		677.86	803)	1.4917	28,955	2591	0,383	766	2,125	0,06258	14,462	0.9036	2402	0950	193.5		
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READING # 0091 BLOCK # 141 TIME # 255,951

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	- •		498)	1,3281	20.84	2119											
10 9.53	296	164.20	234)	1,3816	20.047	1514	2.405	3637	1.618	0.67301	19.864	0.1155	4528	36.045	127.4	٠,	
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577.00	0.157	236	-72	99	1.3936	20.047	752	h. 181	6797	1.759	0.05786	19.869	9005-1	2925	4.140	147.7		
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7 × × ¢ 5	1,2960	1.4990	1.3954	1.5980	1,3729	1.3014	1.3014	1.3729	1.3723	1.3019	1.3020	1.3637	1,3598	1,3120	1.3128	1.2768	1.2834	1,2888
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G	23.615	23.617	23,490	23.497	24.683	22.783	22.824	24.546	25.301	73.034	000	18.971	18,698	18.617	15,526	13.789	13,646	13.840	
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± 4 To a	23,478	23.482	21,538	21,539	21,744	21.945	22.185	22.459	22,332	22.401	22.528	22,571	22,449	22,385	22.315	22,735	22,758	22.748	
4440	1,3050	1,3028	1.3141	1,3180	1.3082	1.2967	1.2976	1.2742	1,2819	1,2788	1.2686	1,2665	1.2737	1.2773	1,2813	1.2582	1,2569	1,2576	
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Reading 92

t = 312.87 sec.

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0.150 TIP NS 11.150 10.219))	074.2(1.6947		5857				*							
11,150	268	960.4		1.3956	28.729	9 20	7.254	4043	POH. 1	0.06113	15.527	6.9917	2906	5.759	191.0		
	3006	674.2	801)	1,2945	20,729	6545	995.0	946	4.120	0.06113	154,61	0.9917	\$ 604		9000		
UNNEL 497.994	900£	0 0		1.2947	28.750					•		•					
991.0 0 011	262		71.	1.3956	50./34	9 4 0	7.211	6509	1.009	0.06531	10.041	1166.0	2010	2.901	190.9		
0 11.150	9005	674.2(801)	1.2945	28.129												
10.600 10.142 INLET THROAT	75.5 75.5	39.550		1.2965	58.729	5269	0.584	986	2.120	0.06331	16.061	0.9917	3010	514.0	190.9		
193,051	2868	632,6(1.2988	28.750	25.59											
9.584 U	1124	145.46	2753	1.3727	50.159	1634	3.021	4457	1.859	0.76612	15.527	0.0791	2573	Se.785	165.7		
THURST CENTRAL SECTION	7 P	0 12 A	7401		28.730	0 2 4 7											
0.105	1060			1.1755		1004	3.114	7667	1.659	0.69648	15.527	0.0000	1957	20.00	4,64		
DENESK		3						•									
104.539	9 1	632.6(760)	1.2989	28,729	5239	i :	•			1				-		
	9 4	0000		1.3015	20.724	2503	. 455	1138	1.451	D#060*7	15.527	0.087	2501	12,319	166.0		
285.409	2639	641.70	746)	1.3015	27.070	2605											
0 10,717	1251	170.20		1.3660	27.070	1772	2.742	4857	1.975	0.76973	15.602	0.0791	2573	58.103	164.9	61.	0.07
0 80.	3				;						•						
050.00%	2675	184.30	805)	1.2997	27.114	2614		4				,					,
		12001		****	****		6.341	70/	7 D		700.61	0.0.786	4362	9/11/6	164.6	0.0	0.16
270.364	2774	635,3(1.3039	27,021	2580			٠.		-						
6,459	1193	160.0(1.3699	150.72	1754	2,787		4633 1.972	0.76093	15.642	9610.0	25.56	5	.602.162.5.	4.19	0.04
961.584	275	7	1701	4004	700	26.13											
10,095	1226	101.5(320)	1997	27.007.5	1757	2.706	4754	1.975	0.75911	15.602	0.0802	251.5	280.46	141.1	0177	0 - 0
TOR O	25					•	•		•								•
	2726	624.6(761)	1.5057	27,005	2560		1	4		:						
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144.873		7	791)	1.3074	25.748	4614											
24696 7.055	1265	186,4(1.3006		1827	2.585	4723	2.089	0.71084	15.606	0.0060	6455	52.169.150	~	C.36.	0.03
155.990	2007	632.00		7605.		80S7											
7.11.7	1220	10000	334)	1.3696	25.705	1798	2,625	4119	2.079	0.71070	15.606	0.0861	2455	54.120	156.7	95.0	10.0
0 X 0	15	~				į											•
154.163	2020	⊸ .	775)	1.3097	25.699	2502				•							
	/ 4 / 1 0 1			1.3046	Z3.04¥	1603	7.604	7 7 7	2.017	0.76902	15.666	0.0663	2422	57.15	150.5	95.0	0.000
86.018	3372	621.0(1.2705	20.480	2843											•
27.396	-	300	743)	1.3064	50.405	2516	1.437	5617	2.170	0.05563	15.000	0.0935	2410	360.08	15412	95.0	£ 4.
20.4	1 7 1 7 27	4 . 4 .	-	3440	9.0	9											ı •
35.725	2961	200000000000000000000000000000000000000	60.00	1.2861	20.7.00	, vo 4	1.5360	1266									;
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¥	2455	2434	2420	2420	2445	4569	2653	2625	2865	80 30 20 20	\$036	3054	507.8	3086	3151	5105	3189	3190	,
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1	15.746	15.746	15,855	15,855	15.855	15,855	15.855	15.855	15,855	15.855	15.890	15.690	0.00.61	068.51	15.840	15.890	5.840	15,890	
4	15454.	.62851	.03140	.03100	.58491	.53758	41067	.39634	.37254	.31751	.26081	.45015	.23581	.23074	.16243	.16861	. 10090 1	.16889	
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MC:L#1	25.194	25.198	25.171	23.174	23.396	23.700	23.978	24.164	23.952	24.127	23.970	24.192	24.014	23.942	23.668	24.409	24,446	25.4047	
GAMPA	1.6727	1.2910	1.2900	1.2899	1.2801	1.2659	1.2520	1.2434	1.2558	1.2468	1.2511	1.2380	1.2492	1.2544	1.2576	1,2239	1.2213	1.1667	
	1063)	1084)	(1041)	1042)	1112)	1209)	(1297)	(1345)	1269)	1321)	1297)	1371)	1307)	1275)	1253) 635)	1432)	1443)	(1639)	
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56.437 4	20	1553	153)	2345	25.411	1267	9.6	5.5.7.8	245.47	76/91-0	35.0	6947.0	3.00	32.0 24.4 2.00 / CE.46 0458	100	3	7.5
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10R		7	34 21					•					•		•		
, m		4768	531.2(1632)	1.1702	25.065	3327				·				-			-
-	5.387	3509		1.2340	25.410	2941	1.857	5459	2.535	96691.0	13.690	0.3050	4200	1240 14,421 201,4 6,65 1,60	201.05	5000	00.1
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۳,	525.	4763	520.7(1630)	1.1708	25.072	3525											
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	916	2894	76.6(1291)	1.2180	25.371	3050	1.541	4701	5.534	0.16549	15.840	4077.0	3178	3178 12.091 200.0 0.85 1.00	2002	20	1.60
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	29.917	47.59	510,9(1621)	1.1704	25.075	5316				,							
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66,491 7	.703	むすつす	149.9(1348) 1.2033 25.328	1.2033	25.328	3092	1.578	4860	2.563	1.578 4880 2.563 0.15385 15.890 0.4032	15.890	0.4032	3214	3214 11.067 202.3 0.85 1.00	204.3	59.0	1.00
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	.917	4759	5	1.1704	25.073	3516							٠	•			
86,667	.61	4586	47.36 7	1.2905	00) 1.2905 25.436	2405	3.033	7284	2.540	2402 3.033 7284 2.540 0.03205 15.640 1.9371	15.040	1.9371	3900	3-625 245-5 0-65 1-00	245.5	0.65	1.00
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	_	4730	1571)	1.1704	25.073	3316											
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7	E G F P	9	~														
~	4.917	4887	625.7(1681)	1,1617	24.437	3364											
	9796	24.70	(794) (104)	1.2636	25.430	5977	2.967	7435	6,563	[764] 1.88888	15.890	1.9371	2993	3993 5.701 251.5 0.65 1.00	451.5	54.0	1.00
9	REGEN	07		•					;		!	•	•	,		•	
~	29.917	4687	7116	1.1617	24.437	3364				81) 1,1617 24,457 3364							
	0.156	1783	"713.4(531)	1.3120	25.436	2138	3.828	2.186	1.564	0.01181	15.000		1 4 6 7	4.1	1-402 247.7 0.85 1 00	2 H C	00.
00	ELSTR.	65					•	•						•	•		2
65	3,051	6065	510.9(1684)	1.1996	25.263	3404						,					
6.451		97.6	1000.66 243)	1.3670	25.430	1510	5.709	8714	2.53/	43 1.3670 25.430 1510 5.789 8714 2.53/ 0.02625 15.490 2.3838	15.490	2.3030	4398	3.555 270.0 0.05 1.00	270.0	0.45	700
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8.687	5	1007	461.1(1601)	1.1745	25.118	3302											
	. 585	4183	#379.0(663) 1.5944 25.436 2350 3.099 7283 2.520 0.u3203 15.890 1.9471	1.2944	55.430	2 \$ 5 0	5.099	7263	2.520	0.03203	048.51		1985	. 3867 . 3.625_244.6 U.85 1.00.	444.0	300	1,00

PEADING # 0042 PLOCK # 216 TIME # 312,867 HACH 7.3 PT # 497,999 TT # 5006.2

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